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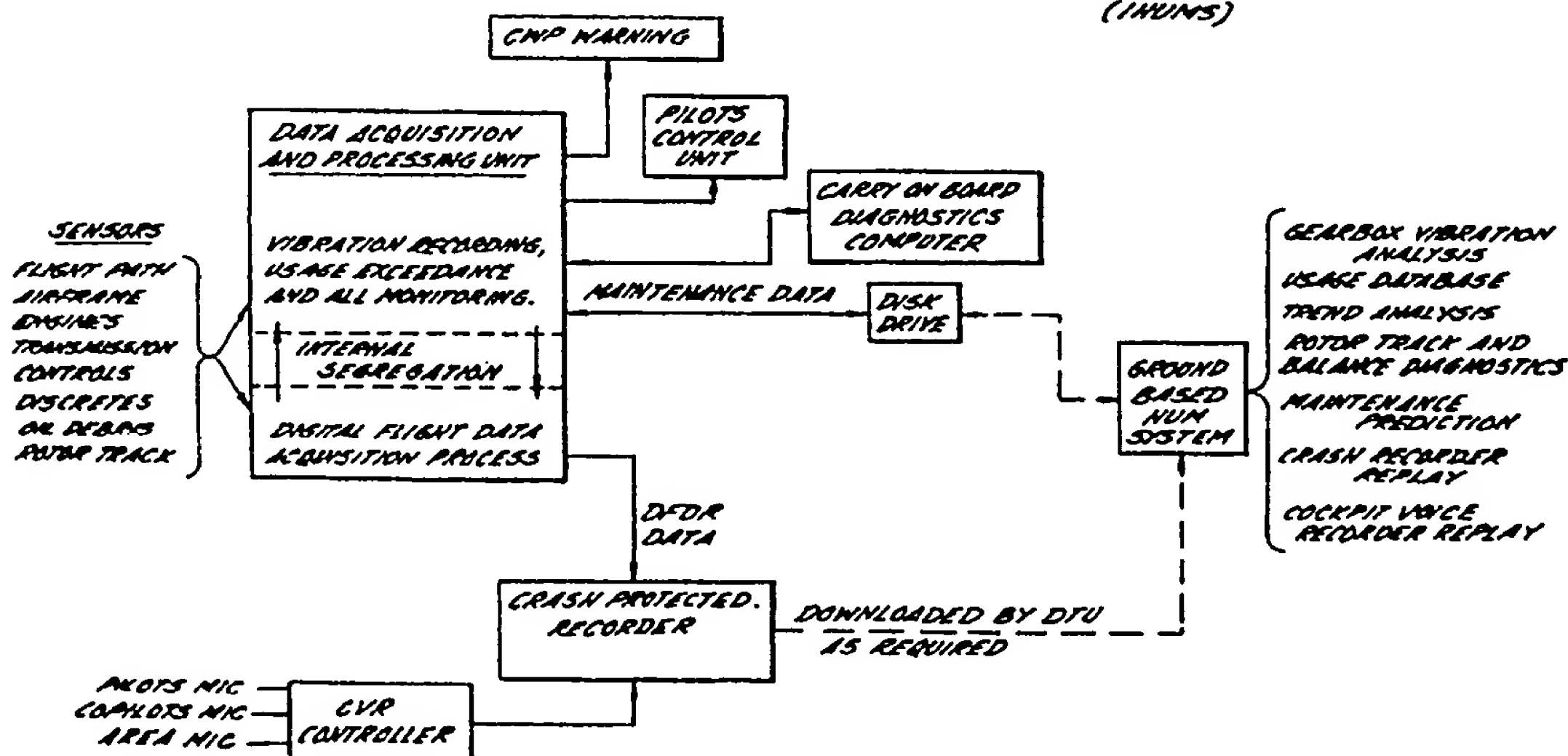
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(54) Aircraft health and usage monitoring systems.

(57) The disclosure relates to an aircraft health and usage monitoring system comprising a computer system (D.A.P.U.) for processing data relating to both flight profiles and selected indicators (SENSORS) of health and usage of an aircraft and means to monitor the flight profile and the selected indicators to supply corresponding data to the computer system. The computer system has predetermined thresholds therein in respect of the selected indicators and is arranged to record the duration and extent by which any threshold is exceeded in use of the aircraft. The indicators of the health and usage of the aircraft are selected from the following: engine health, transmission health, airframe health, aircraft systems health.

*FIG. 1. BUL INTEGRATED FLIGHT  
RECORDING, HEALTH AND  
USAGE MONITORING SYSTEM  
(IHUNS)*



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## AIRCRAFT HEALTH AND USAGE MONITORING SYSTEM

This invention relates to an aircraft health and usage monitoring system. The term "health" as used throughout this specification is intended to mean the condition of a part or system of the aircraft to perform the function required of it. It is known to provide flight data recorders (FDR) and cockpit voice recorders (CVR) in aircraft to record flight data and cockpit crew voices respectively when an aircraft is in use for subsequent analysis in the event of an accident. The object of this invention is to provide a system for monitoring the health and usage of certain important aspects of an aircraft component and system when in use to assist in the maintenance of an aircraft and also to provide relevant data in the event of a failure of a component or system.

Thus the present invention provides an aircraft health and usage monitoring system comprising a computer system for processing data relating to both flight profiles and selected indicators of health and usage of an aircraft, means to monitor the flight profile and the selected indicators and to supply corresponding data to the computer system, the computer system having predetermined thresholds therein in respect of the selected indicators and being arranged to record the duration and extent by which any threshold is exceeded in use of the aircraft.

By way of example, in the case of a helicopter, the selected indicators may include the main rotor track and balance, tail rotor balance, air frame vibration, main gearbox vibration, intermediate gearbox vibration, tail gearbox vibration, accessory gearbox vibration, engine vibration, gearbox oil debris and engine oil debris. More specifically, means may be provided for monitoring and supplying data to the computer system for total engine running times, time spent by engines operating in selected operating bands, time spent by rotor systems including main, intermediate, tail and accessory gearboxes in selected operating conditions and total running time for rotor heads and transmission systems.

The indicators may include means for determining stress on one or more components of the aircraft engine(s)/transmission(s)/air frame and the computer system may include means for calculating low cycle fatigue (LCF) from the stress information obtained in relation to those components and for storing the resulting information for subsequent processing.

The computer system may have output means for writing data accumulated therein from the monitoring means in the course of use of the aircraft to a recording medium for use and analysis by a separate ground based computer system to enable the aircraft health and usage to be monitored.

To enhance flight safety and meet FDR mandatory recording requirements for aircraft, the health and usage monitoring system may be combined with cockpit voice and flight data recorders.

The following is a description of a specific embodiment of the invention, reference being made to the accompanying drawings in which:

Figure 1 is a diagrammatic illustration of an integrated flight recording health and usage monitoring system (HUMS) for an aircraft; and

Figures 2 to 14 illustrate a further embodiment of the invention.

The purpose of the system is to obtain data from various aircraft sensors that will enable the operator to monitor the health of various components of the aircraft establish accurate usage cycles during the aircraft flight profile and to provide aircraft flight data information for the Accident Investigators in the event of an incident occurring on the aircraft.

The system of Figure 1 comprises a Data Acquisition Processing Unit, (DAPU) as illustrated in Figure 14, a Cockpit Display Unit (CDU) as illustrated in Figure 11, a Digital Flight Data Recorder (DFDR), a Maintenance Recorder, a Ground Base Station and/or a carry on board Diagnostics Computer.

The system acquires data, as specified by the CAA, or FAA from the various aircraft sensors which are fed into the DAPU. Examples of such data are listed in the Data Schedule at the end of this description.

The FDR sub system is segregated within the DAPU from the HUM sub system to minimise the risk and consequences of common mode failures. (Normal certification procedures apply to the FDR and CVR parts of the system).

Upon initial introduction of the HUM facilities provided by this combined system, there is no change to the maintenance procedures used to ensure the safety of the aircraft. As data is obtained and confidence in the system established, certain maintenance credits will be developed with the Approval Authorities. All of the airborne software is designed, built and tested to RTCA DO-178A LEVEL II certification procedures.

The signals are processed within the DAPU into four basic categories, Flight Data Acquisition, Exceedances, Health Monitoring and Usage Parameters. The DAPU will interface the Digital Flight Data Recorder (DFDR), a Quick Access Recorder (QAR), a Maintenance Recorder, a Diagnostics Computer and Cockpit Mounted Control Display Unit. The Cockpit Voice signals may be recorded directly onto the DFDR

or onto a dedicated Cockpit Voice Recorder.

In normal operations the DFDR data is only be checked during routine maintenance (normally every 12 months).

The system is designed on a modular basis, so that less extensive systems can be fitted into smaller aircraft. A complete system, such as would be fitted to the larger aircraft as shown in attachment 2 would include:-

(i) COCKPIT VOICE RECORDER (CVR)

Cockpit Voice Recording with separate tracks for Pilots, Co-pilots and Area Microphone Signals on the Combined Accident Data Recorder.

(ii) DIGITAL FLIGHT DATA RECORDER (DFDR)

Flight Data Recording on the Combined Accident Data Recorder with a parameter list which meets or exceeds the CAA, and FAA requirements. Higher data rates are available to accommodate the parameter lists i.e. 64, 128 or 286 x 12 bit words per second (WPS). A proposed parameter list for a large aircraft is shown in attachment 1. Built In Test (BIT) is part of the DAPU and performs testing of the processor and peripherals. The status of the BIT is available on the DFDR data stream.

iii) HEALTH - USAGE TO THE DFDR

This Flight Data Recorder Data Stream will accommodate a limited amount of Health and Usage-Data, covering Rotor Tracks Oil Debris, and detected Exceedance flags.

(iv) DFDR DATA TO THE MAINTENANCE PROCESSOR

The DFDR data system, together with a GMT time and data stamp is copied to the maintenance recorder during the take-off and landing phases to enable the condition of the DFDR sensors and the interface circuitry to be monitored on a daily basis.

(v) CVR/DFDR REPLAY

It is possible to replay the FDR and CVR recordings; safeguards are provided to protect the aircrews' interests with respect to the voice tracks.

(vi) EXCEEDANCE MONITORING

When an Exceedance is detected a report is transferred to the Maintenance Recorder detailing:- the Exceedance Parameter, Duration of Exceedance, Maximum Value of Exceedance, time into flight at the start of Exceedance, and a copy of the FDR data stream extending throughout the period of the exceedance and for a few seconds before and after. An Exceedance flag will also be identified on the FDR data stream to the DFDR. An Exceedance Summary will also be available on the Cockpit Display Unit (CDU), for use by aircrew or maintenance staff.

(vii) LOW CYCLE FATIGUE MONITORING

The DAPU will provide for the Low Cycle Fatigue (LCF) calculations on two engines. The calculations for up to four components on each engine will be calculated by extracting selected flight parameters for each engine from the FDR data stream. The calculated LCF for each component is added to the cumulative count for that component and also stored as its per flight count. The cumulative and per flight LCFs for

each component are maintained in non volatile store. The LCF report for each component is transferred to the maintenance recorder after engine shutdown, together with a GMT Time and Data stamp.

5 (viii) LIFE USAGE MONITORING

10 The DAPU provides Life Usage reports for the two engines and transmission systems comprising of:-1. Total running time, 2. Time spent by each engine above certain speed or torque thresholds, 3. Engine starts, 4. Life Usage reports are also be provided for up to four transmission systems (i.e. Rotors and gearboxes).

(ix) ENGINE PERFORMANCE MONITORING

15 After pilot initiation the DAPU enables the power assurance to be calculated for each engine by extracting the selected flight parameter for each engine from the FDR data stream. The power assurance calculation for each engine is transferred to the Maintenance Recorder and also to the non volatile store for display on the Cockpit Display Unit (CDU) if installed.

20 (x) MAIN AND TAIL ROTOR TRACK AND BALANCE MONITORING

During certain flight conditions the pilot or the system can activate measurements of the main and tail rotor out of balance vibration, and measurements of the main rotor blade track and lag are also obtained.  
25 The DAPU provides a Track and Balance Report comprising of:-1. Main Rotor Track Summary, 2. Main Rotor Balance Summary, 3. Tail Rotor Balance Summary, 4. FDR frames Parameters prior to and after the event. 5. GMT Time and Data stamp which is transferred to the Maintenance Recorder.

30 (xi) AIRFRAME VIBRATION MONITORING

At an appropriate flight condition, the DAPU automatically initiates acquisition of data from the airframe mounted accelerometers and the rotor track sensor. The DAPU airframe vibration report is transferred to the Maintenance Recorder together with a DMT Time, Data stamp and the associated frames of the FDR data  
35 stream.

(xii) GEARBOX VIBRATION MONITORING

40 The DAPU automatically or by Pilot activated switch initiates acquisition of both raw and synchronously sampled and averaged data from accelerometers mounted on the main intermediate and tail gearbox systems. The signals are formed into a gearbox vibration report, together with frames from the associated FDR data, GMT Time and Date stamps and recorded onto the Maintenance Recorder.

45 (xiii) ENGINE VIBRATION MONITORING

The DAPU automatically initiates acquisition of data from accelerometers mounted on the two engines. The DAPU engine vibration reports are transferred to the maintenance recorder together with a GMT Time,  
50 Date stamp and the associated frames of FDR data.

(xiv) OIL DEBRIS MONITORING

55 The DAPU automatically initiates acquisition of data from various types of oil sensors installed in the main, intermediate, and tail gearboxes and, if required, both engines.

The DAPU Oil Debris reports for all components are transferred to the Maintenance Recorder together with GMT Time, Date stamp and the associated frames of FDR data.

(xv) COCKPIT DISPLAY

The system will be capable of driving a full Cockpit Display Unit (CDU) which will provide the pilot with information on exceedances power assurance, alerts etc., together with the command button required for the acquisition of data. If required a simple display of push buttons and lights could be fitted which would enable the pilot to acquire defined data together with advisory/warning lights.

(xvi) MAINTENANCE RECORDER

The Airborne Maintenance Recorder is usually a rugged three and a half inch removable media drive which will interface with the DAPU. This unit has a formatted storage capacity of 1.4Mb. Alternative forms of removable media recorders can be used if required.

(xvii) GROUND STATION

The Ground Station consists of a Personal Computer based system which will include printer and network facilities. The basic Ground Station will be programmed to accommodate diagnostic suites for all aircraft types that are fitted with the HUMS system. The function of the Ground Station will be to:- 1) Produce and Replay disks, 2) Single data entry point for pilots/engineers, 3) Identify defects prior to failure of components, 4) Management of defects and corrective action taken, 5) Indications and remedies for problems found, 6) Provide technical records and management information, 7) Maintenance cycle counts, 8) Provide reliability reports, 9) To generate accurate information and measurements on which engineers can carry out diagnosis of rectification action on aircraft systems. The elements of the Ground Station which relate directly to aircraft maintenance will be developed, built and test to RTCA D0178A LEVEL II certification procedures. The Airborne Maintenance Recorder removable media will be transferred to the Ground Station and will provide Technical Log type information with indications of defects and exceedances that have been identified by the I.HUMS system.

An action items identified will have to be cleared by a Technical Log entry before the aircraft will be cleared for further flights.

(xviii) DIAGNOSTICS COMPUTER

The Computer is used as a front diagnostics line test set to monitor parameters during fault finding and calibration purposes. In addition it also acts as a front line Ground Station for individual aircraft operating away from the main base.

SYSTEM OPERATION

The flying schedule may be entered into the ground station computer (or down loaded from another system) as are the current status of components fitted to the aircraft.

Before each flight the Captain is provided with a removable media containing the basic details of the flight. This removable media is placed in the Airborne Maintenance Recorder on the aircraft, where it confirms that the removable media has been fitted to the correct aircraft.

The aircraft is not be allowed on a further flight until that removable media has been returned to the Ground Station and the aircraft signed off by the Captain and Maintenance Staff. This, in practice, is equivalent to the current Technical Log process.

After completion of the flight, the Captain takes the removable media from the Maintenance Recorder and puts it into the Ground Station Computer. The Short-term Analysis Diagnostic Suites are immediately performed and 1st line reports are printed on the screen together where appropriate with a computer print out. Information displayed are exceedances, component defects, oil debris alerts etc., that have been identified by the airborne system. The report cross-references the action required with the relevant section of the Maintenance Manual.

Rectifying action by the Maintenance Staff can be identified on the Ground Station Computer effectively clearing the Technical Log.



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The Ground Station maintains a log of all faults found. This may be extended by the entry of manually identified snags. These faults can be actioned and closed (or deferred) as required.

Various analysis of the maintenance data base may be performed; for example identify how many servo valves have been changed on one aircraft or how many times gearbox accelerometers have been replaced etc.

The raw data may be analysed using prepared suites of functions for trending and graphing results (i.e. Trending the results of Main gearbox vibration analysis together with the associated oil changes, magnetic plug checks, oil analysis etc.)

The system will provide off line analysis and management summaries as required.

FDR Parameter List for Aircraft (Greater than 2700 Kg)		
Parameter	Sample Rate (Hz) 12-bit Words	Discretes
Time (GMT)		1
Pressure Altitude	1	
Indicated Airspeed	1	
Magnetic Heading	1	
Normal Acceleration	8	
Pitch attitude	4	
Roll attitude	4	
Radio Key		1
No. 1 Engine Ng	1	
No. 1 Engine Nf	1	
No. 1 Engine Q	2	
No. 1 Engine T4	1	
No. 2 Engine Ng	1	
No. 2 Engine Nf	1	
No. 2 Engine Q	2	
No. 2 Engine T4	1	
Main Rotor Speed	2	
Collective Pitch	2	
Longitudinal Cyclic	4	
Lateral Cyclic	4	
Tail Rotor Pedal	2	
Hyd 1 Pressure		1
Hyd 2 Pressure		1
Outside Air Temperature	1	
AFCS engagement Discretes		1

FDR Parameter List for Large Aircraft (Greater than 5700 Kg)		
Parameter	Sample Rate (Hz) 12-bit Words	Discretes
Main gearbox oil pressure	1	
Main gearbox oil temperature	1	
Yaw acceleration	4	
Sling force load	2	
Longitudinal Acceleration	4	
Lateral Acceleration	4	
Radio Altitude	1	
Glideslope Deviation No. 1	1	
Localiser Deviation No. 1	1	
Glideslope Deviation No. 2	1	
Localiser Deviation No. 2	1	
Marker Beam Passage (3 off)		1
Warnings		1
NAV Frequency selection		
DME distance	1	
Latitude	1	
Longitude	1	
Nav computed groundspeed		
Nav computed drift angle		
Landing gear		1

A further example of the integrated Health and Usage Monitoring system (IHUM) in accordance with the invention which will now be described with reference to Figures 2 to 10. The system provides for the automatic sensing and recording of data from a number of strategically positioned sensors, both on the airframe and the major mechanical components; flight Data Recorder parameters are transferred to a combined Cockpit Voice and Flight Data Recorder (CVFDR) whilst the Health and Usage Monitoring parameters are transferred to a Maintenance Data Recorder (MDR).

#### AIRBORNE DIAGNOSTICS

The pilot may interrogate the system during flight, via a Cockpit Display Unit (CDU), and may also enter additional operational information on the CDU, usage and exceedance may be viewed.

Date is transferred onto a removeable media during the flight/mission, for subsequent transfer to a groundstation Personal Computer.

#### GROUND BASED DIAGNOSTICS

Analysis of the data is performed in two stages. The first stage is completed within one minute and identifies the aircraft/component usage, together with any exceedances that may have occurred during the flight which would generate a technical log entry to place the aircraft unserviceable. The second stage takes approximately 3 minutes to complete during which time a large number of parameters are closely analysed.

From some of the data extracted, changes in the state of the helicopter and its equipment may be tabulated. These trends, when evaluated, provide a vital aid to preventative maintenance in that equipment which starts to show an increase in normal levels can be removed before irreparable damage occurs.

#### GLOSSARY OF TERMS

AAIB - Aircraft Accident Investigation Branch  
 ABS - Analogue Digital Converter  
 AIDS - Aircraft Integrated Data System  
 ARINC - Aeronautical Radio Incorporated  
 5 ATA - Air Transport Association of America  
 ATR - Air Transport Radio  
 BALPA - British Airline Pilots Association  
 BHL - Bristow Helicopters Limited  
 BIT - Built In Test  
 10 BITE - Built In Test Equipment  
 CAA - Civil Aviation Authority (United Kingdom)  
 CVR - Cockpit Voice Recorder  
 CDU - Control and Display Unit  
 CVFDR - Combined Voice and Flight Data Recorder  
 15 CRT - Cathode Ray Tube  
 CTS - Clear To Send  
 DAPU - Data Acquisition and Processing Unit  
 DAU - Data Acquisition Unit  
 DEG - Degress  
 20 DFDR - Digital Flight Data Recorder  
 DITS - Digital Information Transfer System (ARINC 429)  
 DOS - Disk Operating System  
 DTU - Data Transfer Unit  
 EEPROM - Electrically Erasable Programmable Read only Memory  
 25 EGT - Exhaust Gas Temperature  
 EPROM - Erasable Programmable Read Only Memory  
 EUROCAE - European Organisation for Civil Aircraft Electronics  
 FAA - Federal Aviation Authority  
 FDAU - Flight Data Acquisition Unit  
 30 FDEP - Flight Data Entry Panel  
 FDR - Flight Data Recorder  
 FPM (fpm) - Feet Per Minute  
 FSK - Frequency Shift Keying  
 GMT - Greenwich Mean Time  
 35 GS - Groundstation  
 G/S - Glide Slope  
 IGB - Intermediate Gear Box  
 IHUMS - Integrated Health and Usage Monitoring System  
 IMP - IHUMS Maintenance Panel  
 40 LCF - Low Cycle Fatigue  
 LRU - Line Replaceable Unit  
 LSB - Least Significant Bit  
 LWC - Liquid Water Content  
 LWL - Left Wing Low  
 45 KN - Knots  
 KIAS - Knots Indicated Airspeed  
 MDR - Maintenance Data Recorder  
 MGB - Main Gear Box  
 MIL-STD - Military Standard (USA)  
 50 MMI - Main Machine Interface  
 MPOG - Minimum Pitch On Ground  
 MRG - Main Rotor Gearbox  
 MSB - Most Significant Bit  
 MS-DOS - Microsoft - Disk Operating System  
 55 MSLS - Multi Sector Load Sheet  
 Nf - Free Power Turbine Speed  
 Ng - Gas Generator Speed  
 Nr - Main Rotor RPM



Nt - Tail Rotor Speed  
 NVR - Non Volatile Ram  
 NVRAM - Non Volatile Random Access Memory  
 ODM - Oil Debris Monitoring  
 5 PAV - Pleassey Avionics  
 PC - Personal Computer  
 PIP - Pilots IHUMS Panel  
 PSD - Project Specification Document  
 Q - Engine Torque  
 10 QAR - Quick Access Recorder  
 QDM - Quantitative Debris Monitor  
 RADS - Rotor Analysis Diagnostic System  
 RAM - Random Access Memory  
 RMS - Root Mean Square  
 15 RNAV - Area Navigation System  
 RPM (rpm) - Revolutions Per Minute  
 RRPM (rrpm) - Rotor Revolutions Per Minute  
 RTB - Rotor Track and Balance  
 RTCA - Radio Technical Commission for Aeronautics  
 20 RTS - Ready To Send  
 RWL - Righth Wing Low  
 RZ - Return to Zero  
 SEC - Seconds  
 SIU - System Interrogation Unit  
 25 SOAD - Spectrographic Oil Analysis Programme  
 SSD - Software Specification Document  
 TGB - Tail Gear Box  
 TRG - Tail Rotor Gearbox  
 TTL - Transistor Transistor Logic  
 30 WHL - Westland Helicopters Limited  
 w.p.s. - Words per second  
 w.r.t. - with respect to

## 1. System Description

Referring firstly to Figure 2, the IHUMS consists of a number of airborne equipments and some ground equipment. An aircraft certification change to accommodate changes in maintenance which may be introduced by IHUMS may be required.

The system is designed to be compatible to all current twin turbine single rotor helicopters. It is not directly applicable to tandem rotor aircraft, although a modified computer unit would be able to provide the dual rotor trackers and larger number of sensor inputs needed for this aircraft.

## 2. System Elements

### A Data Acquisition

The airborne installation is based around a Data Acquisition Processing Unit (DAPU), with an associated Control and Display Unit (CDU) mounted on the interseat console. The DAPU is interchangeable between a number of aircraft types without modification. A considerable amount of extra instrumentation is required, although wherever possible the existing aircraft instruments are used. Several versions of the DAPU are available to meet customers fleet requirements

This unit contains all the conditioning circuitry necessary to sample and accurately monitor a wide range of different types of electrical inputs for subsequent recording, measurement or processing. The mandatory data output interfaces to a standard ARINC 573/717 Flight Data Recorder and a standard ARINC Quick Access Recorder (QAR). Selected mandatory data together with raw and partially processed HUM data is also fed to a Maintenance Data Recorder (MDR).

The DAPU uses twin microcomputers with shared memory to simultaneously perform real time monitoring and intermittent health monitoring tasks. One microcomputer analyses DFDR data in real time for exceedances, flight condition identification, LCF computation etc, thereby leaving the second microcomputer free to handle irregular tasks such as interfacing to the CDU or SIU, and controlling the vibration acquisition tasks. Both microcomputers run the ADA multitasking operating system.

A non-volatile memory is incorporated within the DAPU which contains computed results such as LCF counts, exceedances and component operating times. Documentary data such as a/c ident, engine number, gearbox number etc. is also stored. This memory can be downloaded/uploaded by the aircrew via the Maintenance Data Recorder.

Comprehensive BIT facilities to satisfy the new CAA requirements have also been incorporated.

The Data Acquisition and Processing Unit (DAPU) acquires data for both the Digital Flight Data (accident) and Health and Usage (maintenance) Recorders.

#### 15 B Accident Data Recording

Referring to Figure 3 of the drawings IHUMS adopts a combined voice and data recorder in order to save a very significant amount of weight and space. While this approach is acceptable to the CAA, the design permits any suitable configuration of crash protected recorders to be used albeit with an operational penalty (i.e. the introduction of separate cockpit voice and flight data recorders).

#### C Maintenance Data Recording

Maintenance systems which gather and store data will generate considerable amounts of information, therefore some form of storage is necessary. The solution to these problems adopted by IHUMS is to use electromagnetic removeable media as illustrated diagrammatically in Figure 4.

#### 30 D Ground Station

Ground based computers receive the data carried from the aircraft by the pilot, and carry out analysis and archiving programs. In order that the system can be used to support single operated aircraft as well as large fleet operations, the ground based computer system is able to run on either a modest portable computer or a multi-processor networked system.

#### D Exceedance Monitoring

A large number of exceedance conditions will be continuously available using data from the DFDR data stream and limits and thresholds stored in NVR. The limits and thresholds can be updated, monitored to allow for changes in component modification state. Whenever any one of the monitors detects an exceedance, a time history of the event will be available for ground analysis. This record shall last from a time 4 seconds before the start of the event to a time either 20 seconds later or 4 seconds after the end of the event, whichever is the sooner.

The exceedances are identified using bit mapped words passed into the DFDR data stream and a log record is built identifying the type of exceedance, the duration of that exceedance, the maximum value obtained by the critical parameter and time of the start of the exceedance. For some exceedances it is necessary to establish a cumulative count of the time spent in that condition, and so the DAPU stores a cumulative count in non-volatile form for all monitored exceedances.

#### NOTE:

Some exceedance criteria are subsets of others, for example Ng greater than 105% can only occur while Ng is greater than 100%. In cases such as this, one time history will be created, relating to the exceedance Ng greater than 100%, yet two log records will be created to identify the time spent in exceedance of each threshold level separately.

E Low Cycle Fatigue

Low Cycle Fatigue (LCF) indices are calculated for up to four components on each engine. The algorithms used are obtained from the Aircraft Operating Manual or Engine Manufacturers manual. The incremental and cumulative LCF count are held in NVR, upload along with other component lives and cycle counts. To support accurate LCF calculations the DAPU accommodates Ng sampling at 4 Hz and Nf sampling at 2 Hz.

F Engine Performance Monitoring

The DAPU implements the engine power assurance algorithms from the aircraft manufacturers flight manual. Wherever possible this will permit preflight assurance checking.

It is often necessary for the pilot to ensure that the engines are set in a suitable operating condition before gathering data for power assurance purposes, therefore this process will always be pilot initiated. On some aircraft a single initiation will relate to both engines, whereas for other aircraft types the engines are checked separately and so two initiations are required. In either case, whenever the pilot initiates engine performance monitoring, a frame of DFDR data shall be stored onto the maintenance recorder.

The DAPU will compute the performance indices using algorithms pertinent to that aircraft type.

The result shall be computed in not more than 30 seconds and preferably in less than 10 seconds. A data file containing a DFDR data frame will still be stored to allow the ground computer to obtain power assurance data for trending and any other analysis.

G Trending Data

This facility allows the pilot to enter data into the IHUM system via the CDU, normally when the aircraft is flying in the cruise. Such data is used for trend monitoring over a number of flights, and includes parameters which are not instrumented by the DAPU such as hydraulics and engine oil temperatures and pressures.

When the pilot has entered this data, a frame of DFDR data is added to identify the flight condition and to allow all the instrumented parameters to be trended over consistent flight conditions.

H Track and Balance

Track and balance data capture falls into three categories. Firstly, data is automatically acquired on routine flights (in a number of flight conditions on each flight) from which the health of the main and tail rotor systems are monitored using analysis programs running on the ground computer. Secondly, the pilot can initiate data capture to identify particular problems on the aircraft when necessary, and thirdly it is possible to monitor the airframe vibration and main rotor track signals using the SIU during maintenance test flights.

(1) Routine Acquisition for On Board Processing

On every flight, a measurement of the main and tail rotor out of balance vibration and main rotor track is made at a number of positions around the aircraft, and data is recorded automatically during identified flight conditions so that the IHUM groundstation can develop a trend of rotor order vibration magnitude. When the vibration exceeds a threshold level, or when the corrective maintenance action is sufficient to warrant implementation the groundcrew are prompted to carry out a corrective maintenance action. When this has been carried out the aircraft will be returned to service. A complete set of rotor track and balance data is then gathered on the next flight to check that the maintenance action has been applied correctly and to check that the level of vibration has decreased. The two data sets (obtained before and after maintenance) then provide a basis for refinement of the maintenance algorithms.

(a) Main Rotor Track

The system interfaces directly to either an active or a passive rotor track sensor as illustrated in Figure 5. Data from the sensor is used both for acquisition of data for diagnostic purposes (track and balance) and for recording blade position information for the DFDR.

#### (b) Main Rotor Balance

Up to eight accelerometers are fitted around the cockpit, cabin and tail to measure the airframe vibration due to main and tail rotor out of balance and aerodynamic forces.

#### (c) Tail Rotor Balance

The tail mounted accelerometer(s) is used to monitor tail rotor out of balance forces, using signal processing as for the main rotor forces except that the processing is synchronous to a tail rotor azimuth marker as illustrated in Figure 6.

#### (d) Airframe Vibration

Data is obtained from the airframe accelerometers for diagnosis of vibration at other than the first harmonic.

#### (2) Pilot Initiated Acquisition

The data acquired is the same as for routine acquisition, except that it is annotated as having been required by the pilot.

#### (3) Maintenance Acquisition

During dedicated maintenance test flights, the SIU can be carried onto the aircraft and connected into the IMP port. The maintenance engineer is then able to initiate any of the DAPU acquisition modes, with the data being fed directly back to the SIU.

#### I Gearbox Vibration

Gearbox vibration data is used to detect gear and/or bearing defects within all of the stages of the helicopter gearboxes including the accessory drives. Data capture may be initiated by the pilot but is normally be acquired automatically for on board processing to monitor airborne capability. The downloaded data is analysed using special-to-type programs on the ground to yield information on the condition of each element of each gearbox in the aircraft. Automatic recognition of alert levels in the groundstation is essential because of the large number of parameters present.

#### J Engine Vibration

Two types of engine vibration monitoring are supported, namely in-flight and ground based monitoring.

#### (1) In-Flight Monitoring

Due to the rapid propagation rate of certain gas turbine failure modes, limited continuous monitoring is performed by the DAPU as illustrated in Figure 7.

The basic sequence of operations runs as a continuously repeating loop having lower priority than any pilot, maintenance crew or automatically initiated task. The loop monitors each of the four engine shaft in

sequence before repeating the cycle. While no other operations are in progress the loop cycle time is approximately one minute. This time is extended if other tasks which use the same signal paths are in progress.

5

## (2) Routine Monitoring

Routine engine vibration analysis is based upon recording both the raw vibration signals from the engine mounted accelerometers and the engine speed parameters fed to the FDAU. On the ground, spectral analysis techniques are used to compute a vibration spectrum and the speeds of the rotating components are incorporated to identify any unwanted peaks which occur.

## K Ancillary Equipment Vibration

15

A single piece of ancillary equipment such as an oil cooler fan can be monitored using an accelerometer and, if necessary, a dedicated magnetic pickup for synchronisation. Synchronous or asynchronous sampling will be available and the data acquisition processes described in Appendix B for gearbox vibration will be used.

20

## L Oil Debris Monitoring

On-line oil debris monitoring will be included when suitable sensors become available.

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## M Oil Sampling

IHUM as a system supports oil sampling by prompting maintenance personnel to take oil samples at the required periodicity, and by accepting oil analysis results into the IHUM groundstation database. It will be possible to recall and plot concentrated oil debris records over the life of any gearbox or engine. Data relating to component changes and oil changes will be presented alongside oil debris results.

## N Configuration Control

The flight data acquisition section of the DAPU is pin programmable between eight aircraft types for IHUMS. However, there is an additional requirement for certain categories of data, such as exceedance limits, to be modified in the field. This type of data is held in non-volatile storage thereby enabling the system to monitor and record any exceedances correctly. To control parameter changes in non-volatile storage, configuration control codes in that storage are changed whenever the controlling parameters are modified. A test of the configuration control is carried out on every start up by comparing the expected control codes included in the MDR removable media uploaded data with those already in NVR. This method allows a DAPU to be switched between aircraft types and to monitor exceedances and record the data that the preset parameters for that aircraft type dictates.

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## O Operational Data Input

The pilot is able to initiate IHUMS processes and enter operational data, on the ground and in-flight, which is stored on the MDR for downloading into an operational system.

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## Component Descriptions

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## A Data Acquisition and Processing Unit (DAPU)



This unit contains all the conditioning circuitry necessary to sample and accurately monitor a wide range of different types of electrical inputs for subsequent recording, measurement or processing. The mandatory data output interfaces to a standard ARINC 573/717 Flight Data Recorder and a standard ARINC Quick Access Recorder (QAR). Selected mandatory data together with raw and partially processed HUM data is also fed to a Maintenance Data Recorder (MDR).

The DAPU uses twin microcomputers with shared memory to simultaneously perform real time monitoring and intermittent health monitoring tasks. One microcomputer analyses DFDR data in real time for exceedances, flight condition identification, LCF computation etc, thereby leaving the second microcomputer free to handle irregular tasks such as interfacing to the CDU or SIU, and controlling the vibration acquisition tasks. Both microcomputers run the ADA multitasking operating system.

A non-volatile memory is incorporated within the DAPU which contains computer results such as LCF counts, exceedances and component operating times. Documentary data such as a/c ident, engine number, gearbox number etc. is also stored. This memory can be downloaded/uploaded by the aircrew via the Maintenance Data Recorder.

Comprehensive BIT facilities to satisfy the new CAA requirements have also been incorporated.

The Data Acquisition and Processing Unit (DAPU) acquires data for both the Digital Flight Data (accident) and Health and Usage (maintenance) Recorders.

The functions of the DAPU are as follows:

#### Usage (Maintenance)

(a) It will record and retain in non-volatile memory the Engine Running Time for each engine, the gearboxes and other lifted components.

(b) The DAPU will calculate and retain in non-volatile memory Cumulative counts for engine starts, rotor engagement, take-off and landings, perform exceedance monitoring of engine, rotor and transmission parameters. Exceedance recordings include a complete time history from the FDR data stream etc. for post flight analysis and a summary for display on the cockpit CDU.

(c) Low Cycle Fatigue indices will be calculated for up to four components on each engine.

(d) Engine health monitoring will be calculated automatically on pilot demand.

#### Health (Maintenance) On Board Processing

The DAPU provides the following "in flight" diagnostics:-

(a) continuously perform exceedance monitoring of engine, rotor and transmission parameters. Exceedance recordings will include a complete time history from the FDR Data Stream etc. for post-flight analysis, and a summary for display on the CDU. Exceedances will be classified to yield outputs suitable for inclusion in the DFDR data stream (36 flags) or for the PIP (2 levels of annunciation).

(b) monitor the engines for excessive vibration.

(c) perform engine exhaust gas monitoring when suitable sensors become available.

(d) upon pilot initiation, the system will:-

(1) sequentially gather data for:-

(a) airframe vibration.

(b) vibration of main, intermediate, tail, and where fitted, combiner gearboxes.

(c) vibration of ancillary components.

(2) sequentially gather data for main rotor track and balance and tail rotor balance.

(3) carry out engine performance monitoring when demanded by the pilot.

(4) enter trending data.

(5) display system status on CDU.

(e) If no pilot initiated task is in progress, the SIU may be used to initiate any one of the following types of data acquisition:-

(1) airframe vibration.

(2) main rotor track and balance.

(3) tail rotor balance.

(4) vibration of the main, intermediate, tail, accessory and, where fitted, combiner gearboxes.

(5) engine vibration.

(6) ancillary equipment vibration.

(7) oil debris sensor monitoring (when sensors are available)



( 8 ) engine exhaust gas monitoring (when sensors are available).

( 9 ) DFDR monitoring.

NÖTE: When initiated from the maintenance computer these process output data only to the SIU and not to the maintenance recorder. (f) If neither pilot or maintenance engineer initiated tasks are in progress, the DAPU will detect certain specified flight conditions and at the correct condition automatically initiate the appropriate process from the following:-

( 1 ) recording FDR data to the maintenance recorder.

( 2 ) airframe vibration, vibration of the main, intermediate, tail, accessory and, where fitted, combined gearboxes, engine vibration and ancillary equipment vibration.

( 3 ) main rotor track and balance and tail rotor balance.

NÖTE: Any automatically initiated data gathering process will be nullified by a manually initiated process.

#### (4) Self Test

The DAPU performs continuous self-testing.

#### Control and Display

Types of pilot display utilised are:-

##### (1) Pilots IHUMS Panel (PIP)

##### (a) PIP Description

Referring to Figure 8 of the drawings, the PIP is a small panel of ARINC rail width, with all of the controls and indications necessary to use the IHUM system. However, it is not able to perform the functions which require a test output and so exceedance display, power assurance results display, operations data entry etc. are not catered for in systems fitted with this panel.

PIP units are to be installed on small aircrafts such as Bell 212, S76 etc. where space is limited, associated with an IMP unit the panel of which is illustrated in Figure 12.

PIP panel is provided with background lighting to illuminated legends. Power is 28V dimmed.

##### (b) PIP Buttons

The PIP illustrated in Figure 13 has six press button momentary action switches. Of these, five are associated with the IHUM DAPU functions, and carry legends that are illuminated while the function is active.

These switches are inscribed:-

TRK/BAL Green

EVENT: Green

CREW CHG: Amber

VIBRATION: Green

PWR ASS: Green

The other switch is a basic unlit push button which enables to perform a lamp test and is marked "LAMP-TEST".

##### (c) PIP Indicators

PIP panel is provided with two dual lamp sunlight readable square indicators to reflect Engine 1 and 2 vibration warnings and MDR warning.

These warning captions are marked

- ENG VIB

- MDR

Illuminated switches bulbs and the ENG VIB and MDR indicators are powered by the 28V Day/Night.

DATA SUMMARY	
POWER (LAMPS)	28V DC TO DO-160B
POWER (LOGICS)	5V DC (DERIVED FROM 15V DC)
POWER OPERATING	15V DC
ENVIRONMENTAL:	TO RTCA DO-160B
TEMPERATURE:	CATEGORY A1
ALTITUDE/HUMIDITY:	OPERATING 15 DEGREES C TO 70 DEGREES C
VIBRATION/MACH SHOCK:	CATEGORY J/Y
EMC COMPATIBILITY:	CATEGORY A

#### (d) Use of DAPU-Related Functions

##### ( 1 ) Event

Pressing the EVENT button will activate the pilot event function, and for a period of 5 seconds from that moment the caption will be illuminated. This function shall operate at all times, including a crew change.

##### (2) Vibration Acquire

Pressing the VIBRATION button will initiate a complete set of vibration acquisitions. During this period the caption will be illuminated. This function may only be selected if the TRK/BAL and CREW CHG captions are extinguished.

##### (3) Acquire Rotor Track and Balance Data

When the pilot presses the TRK/BAL button, the IHUM DAPU will start to acquire the data as defined by the vibration acquisition table for pilot initiated track and balance. While the acquisition is in progress, the caption shall be illuminated. This function may only be selected if the VIBRATION and CREW CHG captions are extinguished.

##### (4) Power Assurance

Pressing the PWR ASS button will initiate a power assurance data acquisition, and the caption will be illuminated for a period of 5 seconds. This function may only be selected if the CREW CHG caption is extinguished.

##### (5) Crew Change

This option is used to initiate the rotors running crew change option. If the pilot presses this in flight, nothing will happen and the caption will not illuminate. Similarly, the option shall not operate when any other caption (TRK/BAL, VIBRATION, EVENT or PWR ASS) is lit.

If the aircraft is on the ground and no caption is lit when this option is selected, the caption will illuminate and the DAPU will write the appropriate data files to the MDR. The pilot may extract the removable media from the MDR when the "Busy" lamp located inside the door of the MDR goes out.

When a removable media for the next crew has been prepared by the groundstation and inserted into the MDR, a second press of the crew CHG button will instruct the DAPU to read the required data files from the MDR. The caption will extinguish on the second press.

(2) Cockpit Display Unit (CDU)

5 The CDU illustrated in Figures 9 and 11 is shared with the RNAV navigation system, having both ARINC 429 and RS422 interfaces. In some cases the CDU will be dedicated only to IHUMS, although both interfaces will still be fitted.

10 This unit acts as the centralised acquisition and entry panel for all forms of data. The CDU is the Racal RNAV2 having a monochrome CRT display. The unit will have dual capability being able to act as both an RNAV2 navigational computer, and a HUM CDU. This gives the advantage that on all aircraft fitted with an RNAV2 installation, a single display will be available, thus further reducing the weight and size of two separate units.

(a) Control of Operating Mode

15 Dedicated IHUMS only CDU.

In this case the CDU shall power up in IHUM Mode, and will stay in that mode in all cases.

(3) Shared RNAV-2/IHUM CDU

20 In this application illustrated in Figure 10 the CDU will power up in RNAV-2 mode, and may be switched back and forth between RNAV-2 and IHUM modes under pilot control as defined in this paragraph.

To obtain access to the IHUMS system from RNAV-2, the "up" and "down" keys will be pressed together. On change of modes the CDU will transmit a "DATA" key code to cause the IHUM system to write the main IHUM menu screen.

25 To obtain access to the RNAV system the pilot may (a) press the "up" and "down" keys together, in which case the system will display the last screen used before leaving RNAV-2, or (b) press one of the specific keys (NAV, VNAV, NAV AID, Flight PLAN or Go Direct) to bring up the associated screen directly, or (c) press DATA, when in IHUM Mode with an RNAV warning caption shown in the top right of the screen, in which case the CDU will change to RNAV mode and bring up the warns page directly.

**Claims**

- 35 1. An aircraft health and usage monitoring system comprising a computer system for processing data relating to both flight profiles and selected indicators of health and usage of an aircraft, means to monitor the flight profile and the selected indicators and to supply corresponding data to the computer system, the computer system having predetermined thresholds therein in respect of the selected indicators and being arranged to record the duration and extent by which any threshold is exceeded in use of the aircraft, the
- 40 indicators of the health and usage of the aircraft being selected from the following: engine healthy transmission health, airframe health, aircraft systems health.
2. An aircraft monitoring system as claimed in Claim 1, wherein the computer system includes means for downloading information recorded in the computer system relating to the aircraft health and usage for on board processing or subsequent analysis.
- 45 3. An aircraft monitoring system as claimed in Claim 2, wherein the computer system has output means for writing data accumulated in the computer system from the monitoring means in the course of use of the aircraft to a recording medium for on board processing or use and analysis by a separate ground based computer system to enable the aircraft health and usage to be monitored.
4. An aircraft monitoring system as claimed in any of Claims 1 to 3, including display means for displaying
- 50 to the aircrew when one or more of the selected indicators is exceeded.
5. An aircraft monitoring system as claimed in any of the preceding claims and in the case where the aircraft is a helicopter wherein the indicators are selected from the following: main rotor track and balance; tail rotor balance; airframe vibration; main gearbox vibration; intermediate gearbox vibration, tail gearbox vibration; accessory gearbox vibration; engine vibration; gearbox oil debris and engine oil debris.
- 55 6. An aircraft monitoring system as claimed in Claim 4, wherein means are provided for monitoring and supplying data to the computer system for total engine running times, time spent by engines operating in selected operating bands; time spent by rotor systems including main, intermediate, tail and accessory gearboxes in selected operating condition and total running time for rotor heads and transmission systems.

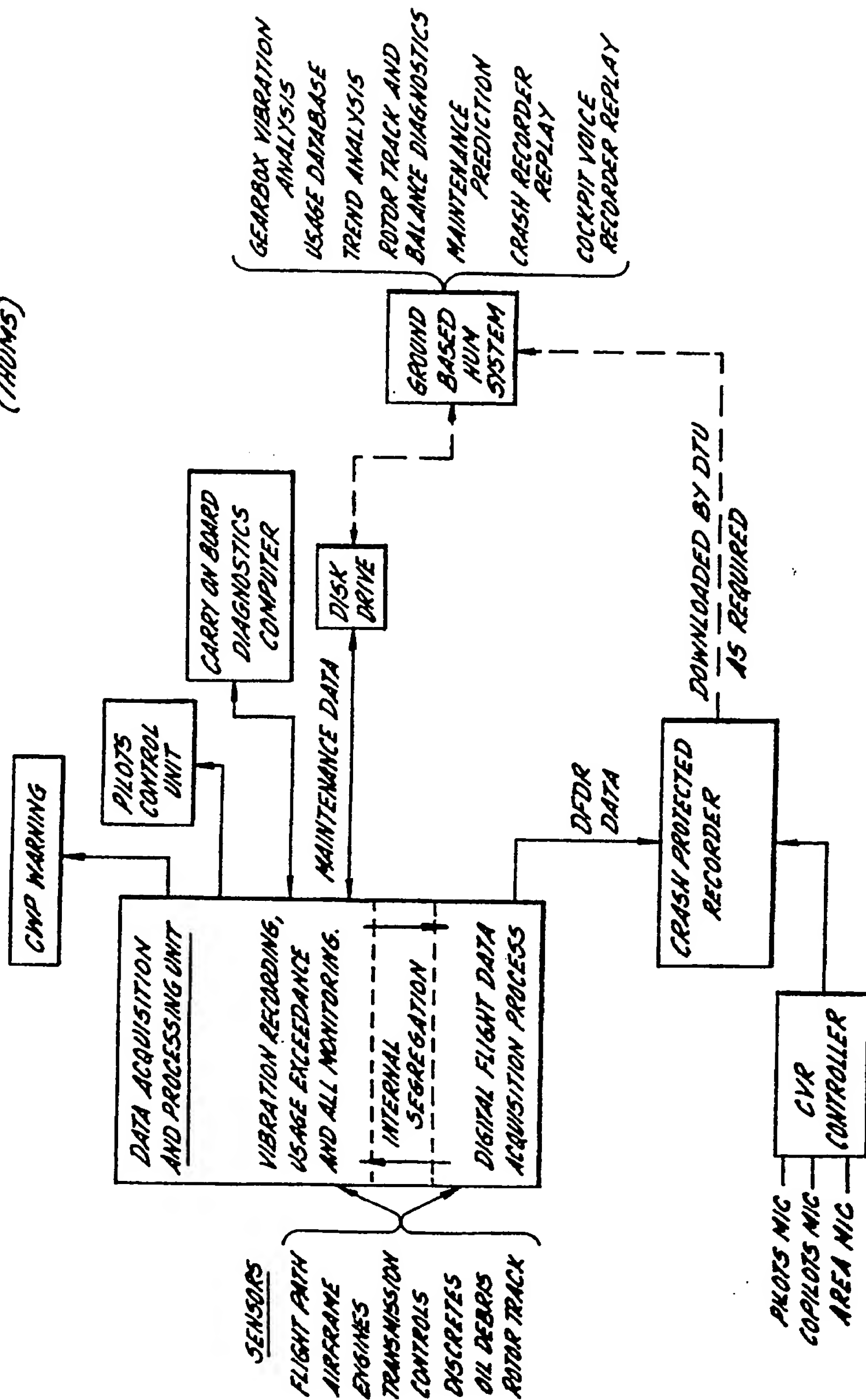
7. An aircraft monitoring system as claimed in Claim 5 or Claim 6, wherein the indicators include means for determining stress on one or more components of the aircraft engine(s)/transmission(s)/air frame and the computer system includes means for calculating low cycle fatigue (LCF) from the stress information obtained in relation to those components and for storing the resulting information for subsequent processing.

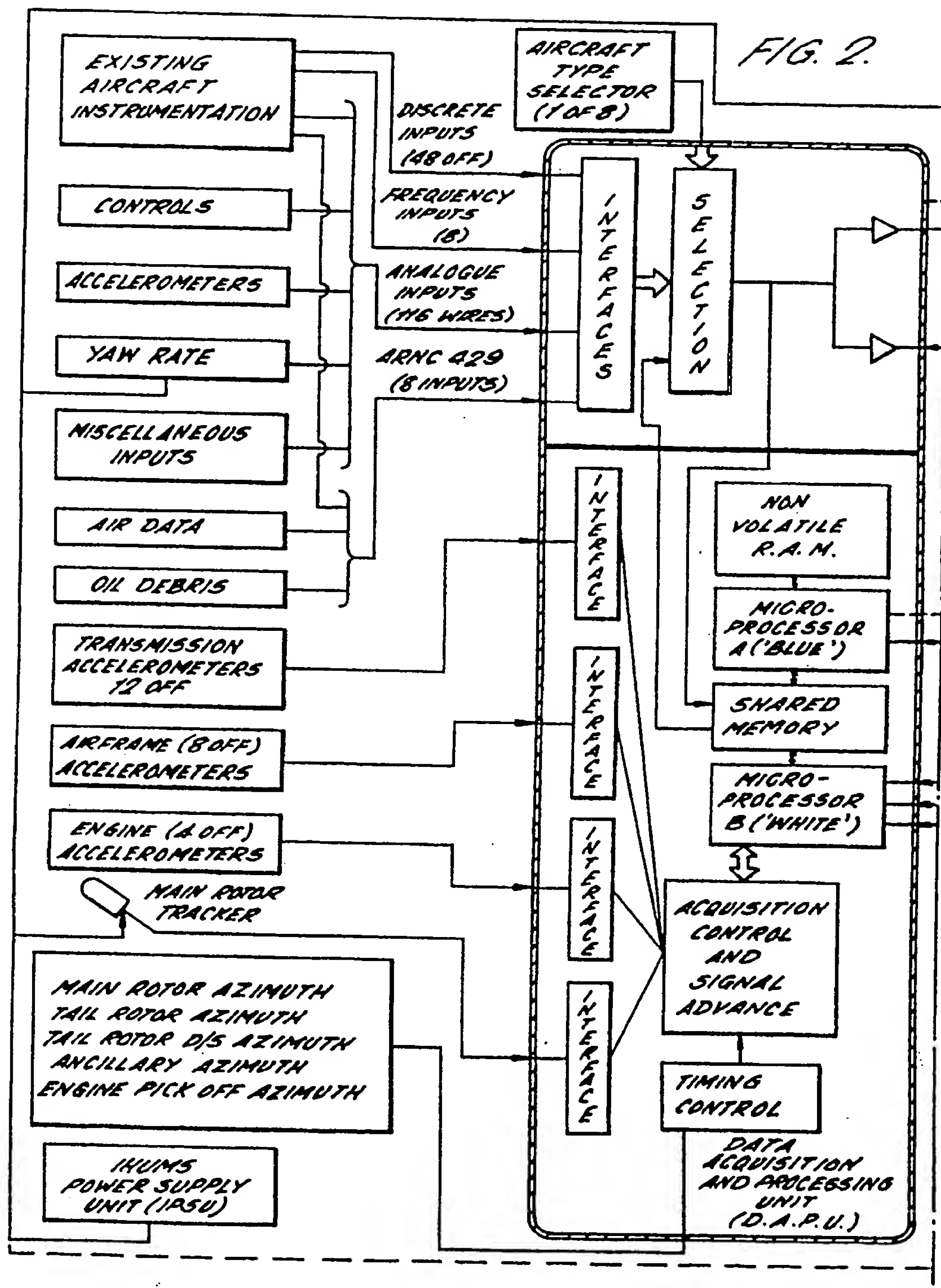
8. An aircraft system as claimed in any of the preceding claims wherein means are provided for supplying information relating to engine performance to the computer system and the computer system includes means for calculating the engine performance and storing the resulting information for subsequent processing.

9. An aircraft monitoring system as claimed in any of the preceding claims wherein a crash protected recorder is provided which is coupled to the computer system to receive selected flight data and health information therefrom.

10. An aircraft monitoring system as claimed in any of the preceding claims, wherein a cockpit voice recorder is included in the system.

FIG. 1. BHL INTEGRATED FLIGHT RECORDING, HEALTH AND USAGE MONITORING SYSTEM (IHUMS)







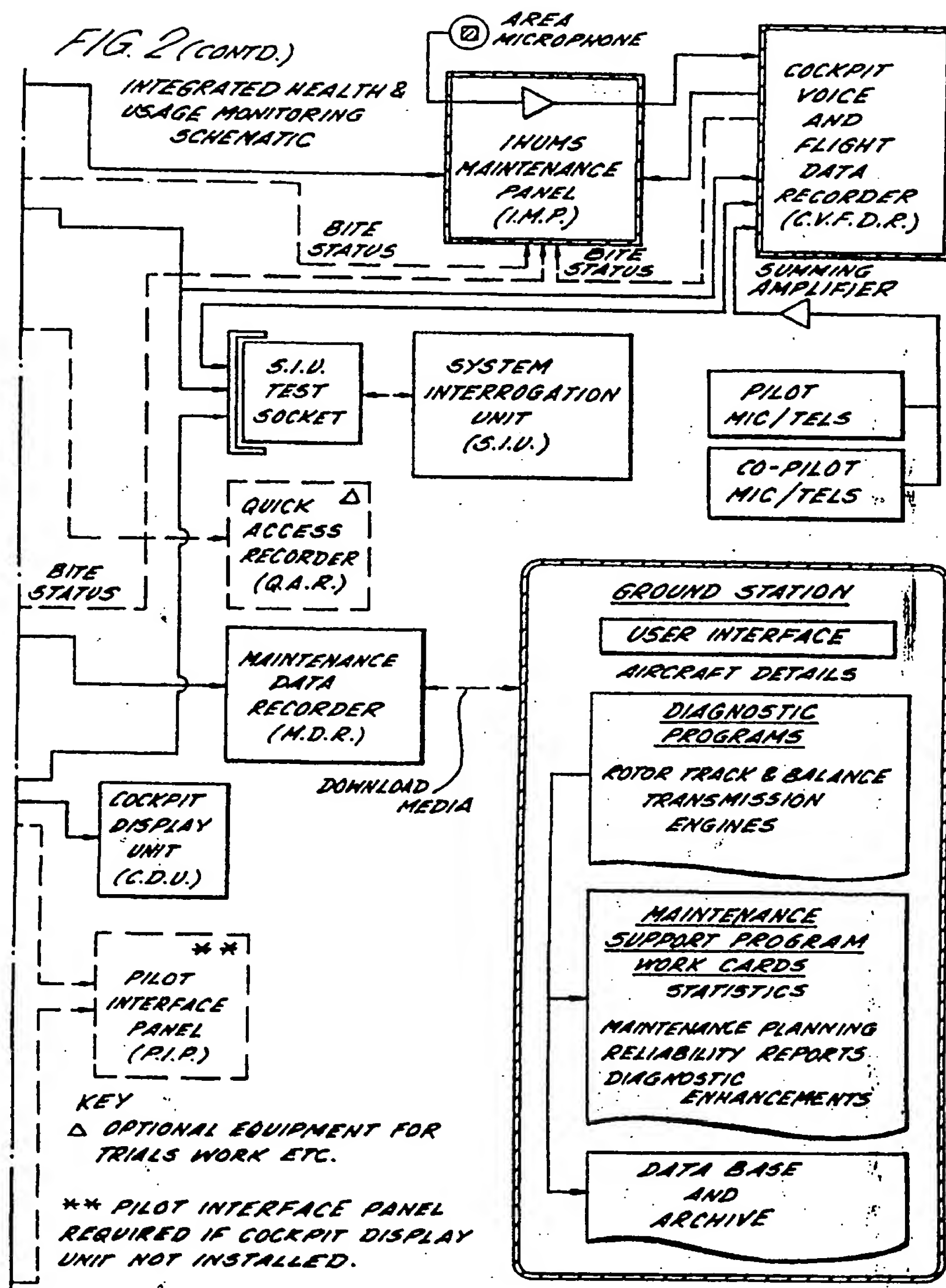


FIG 4

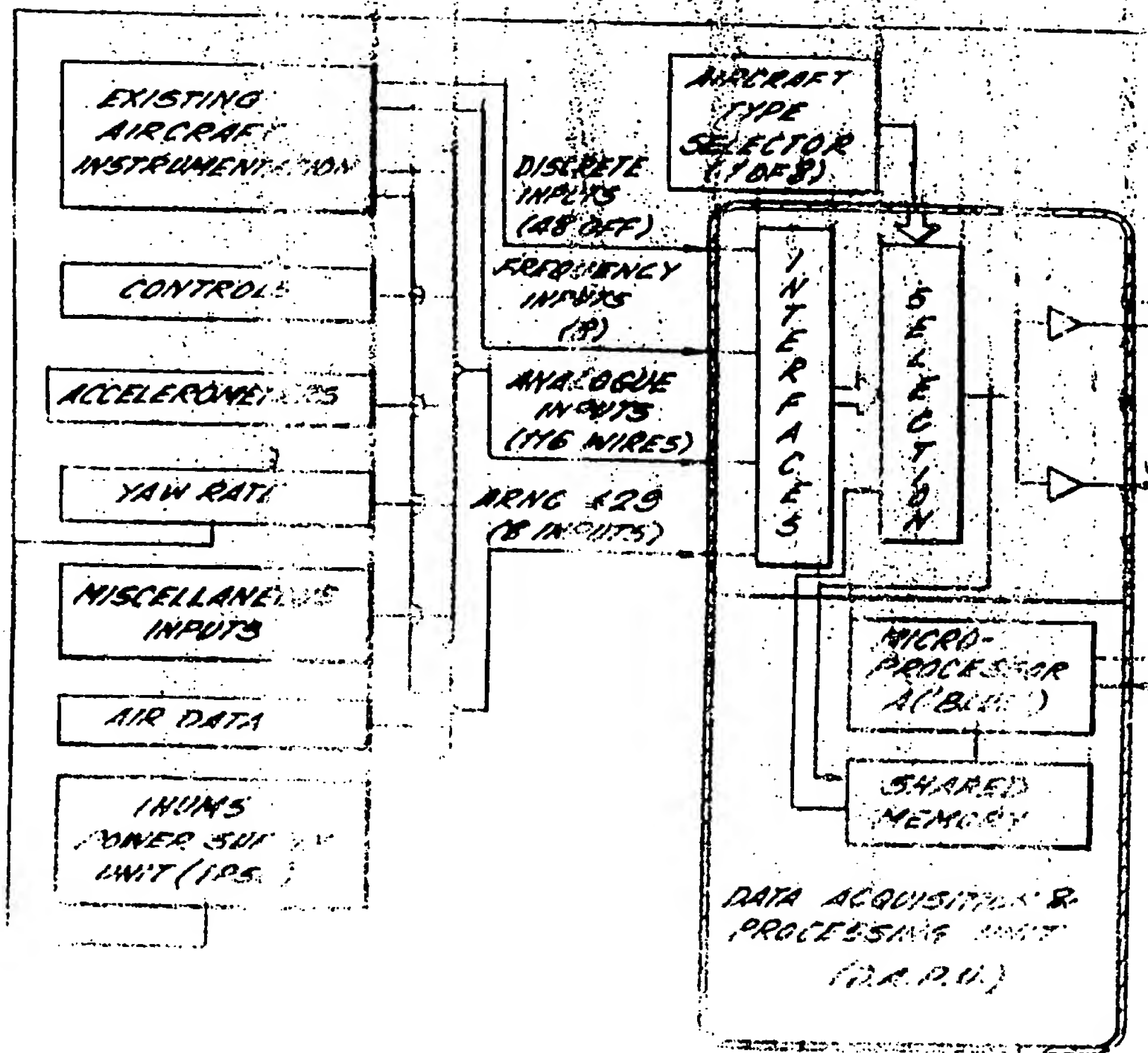
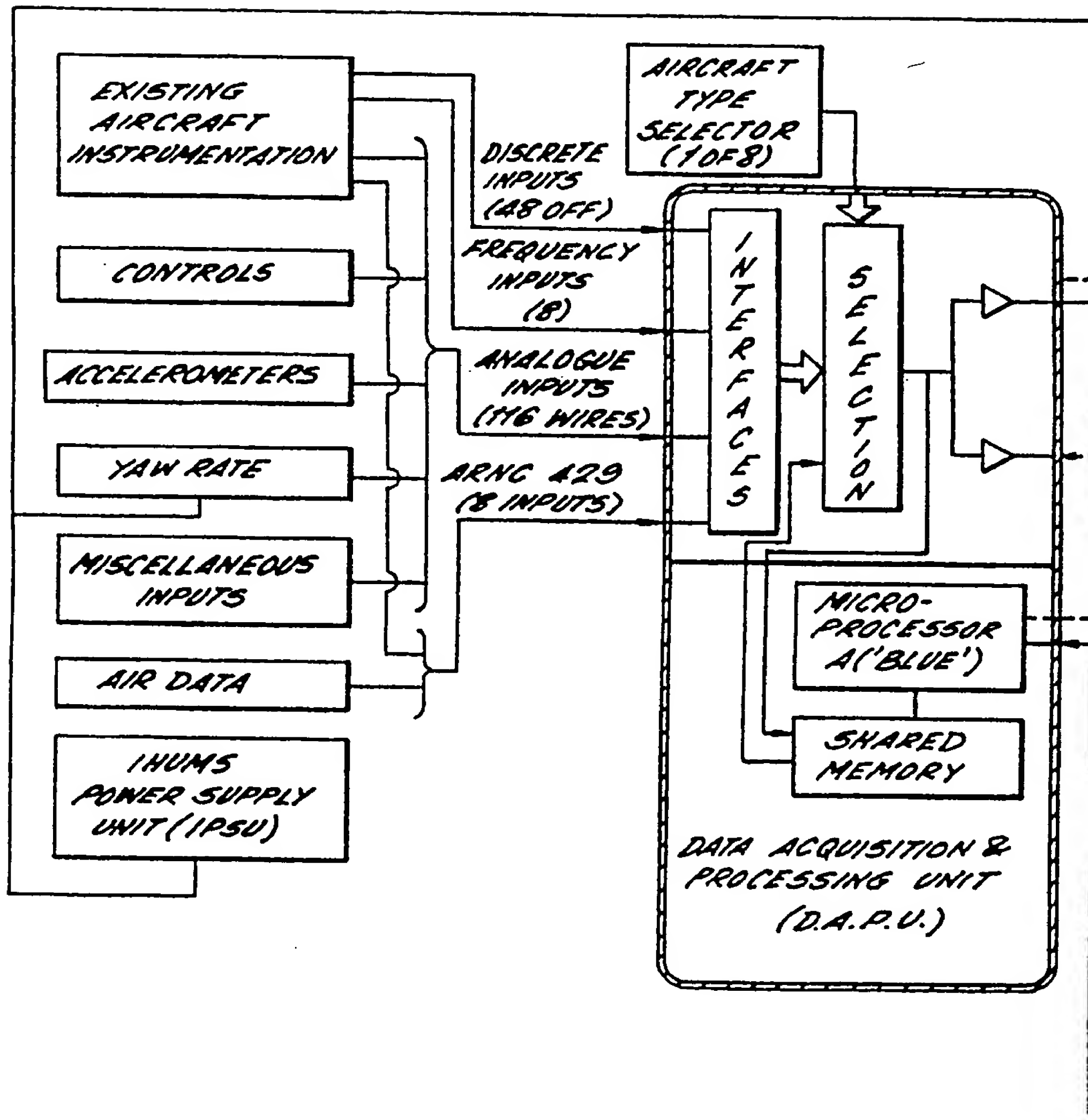


FIG. 4.



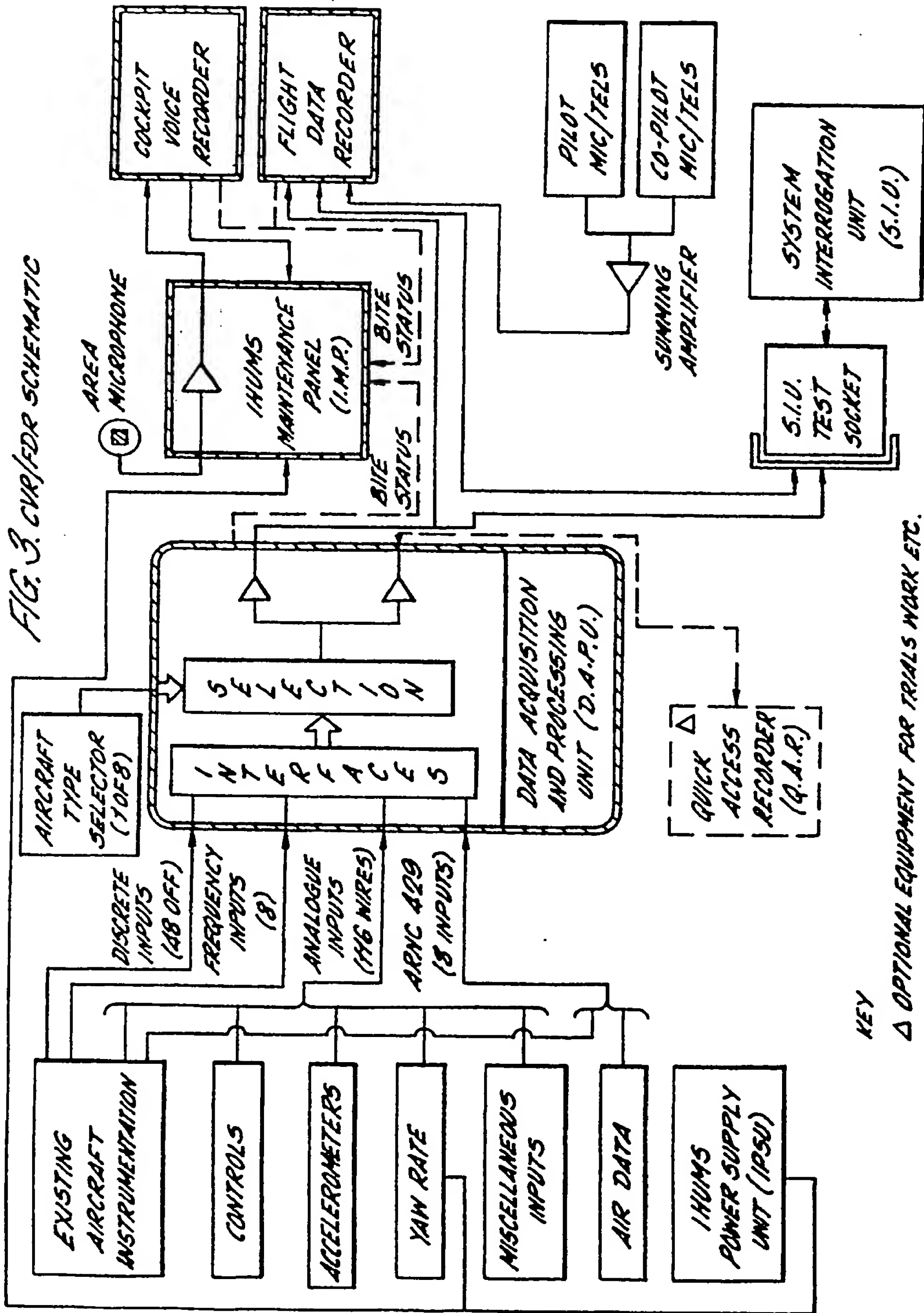
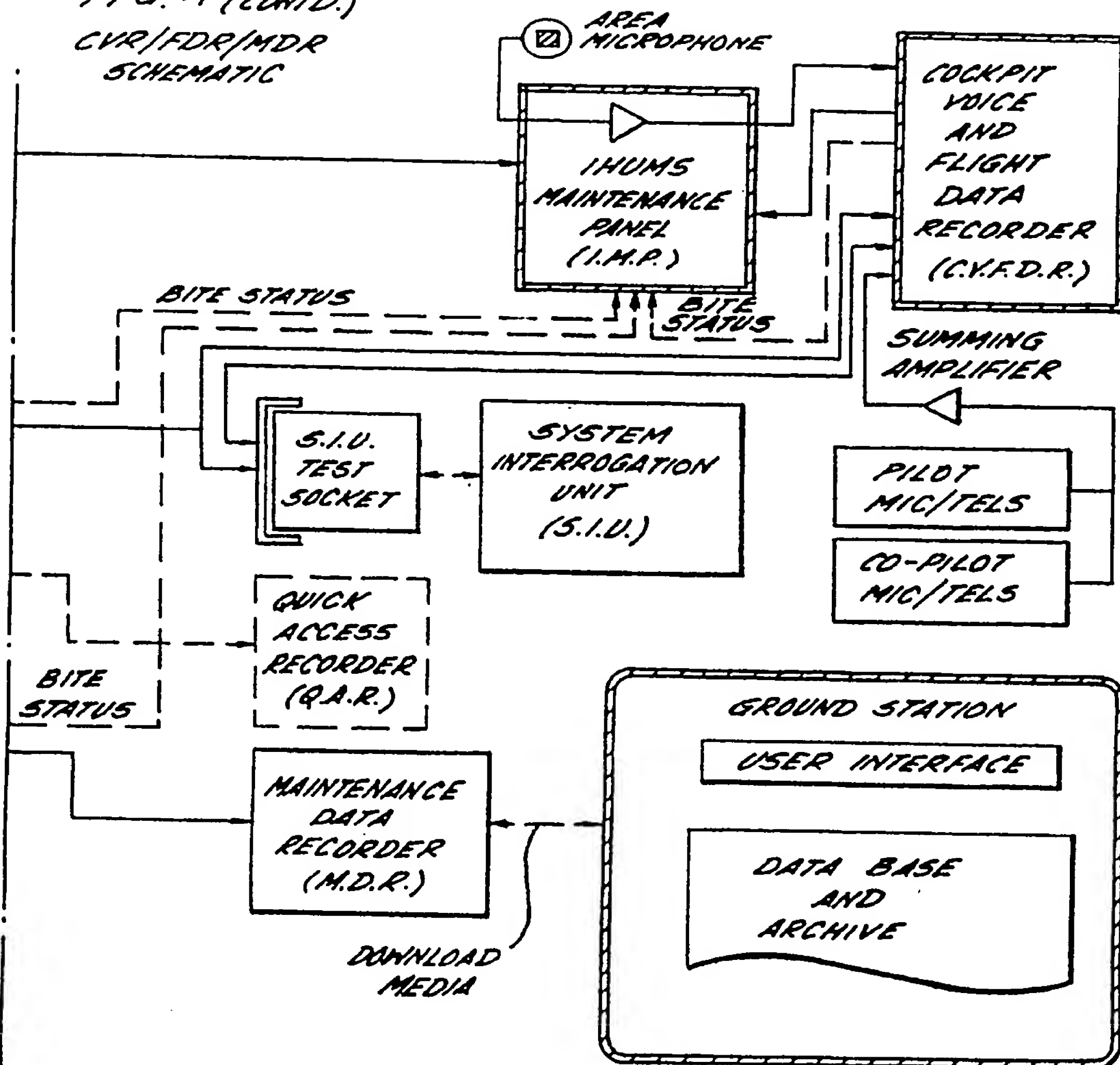
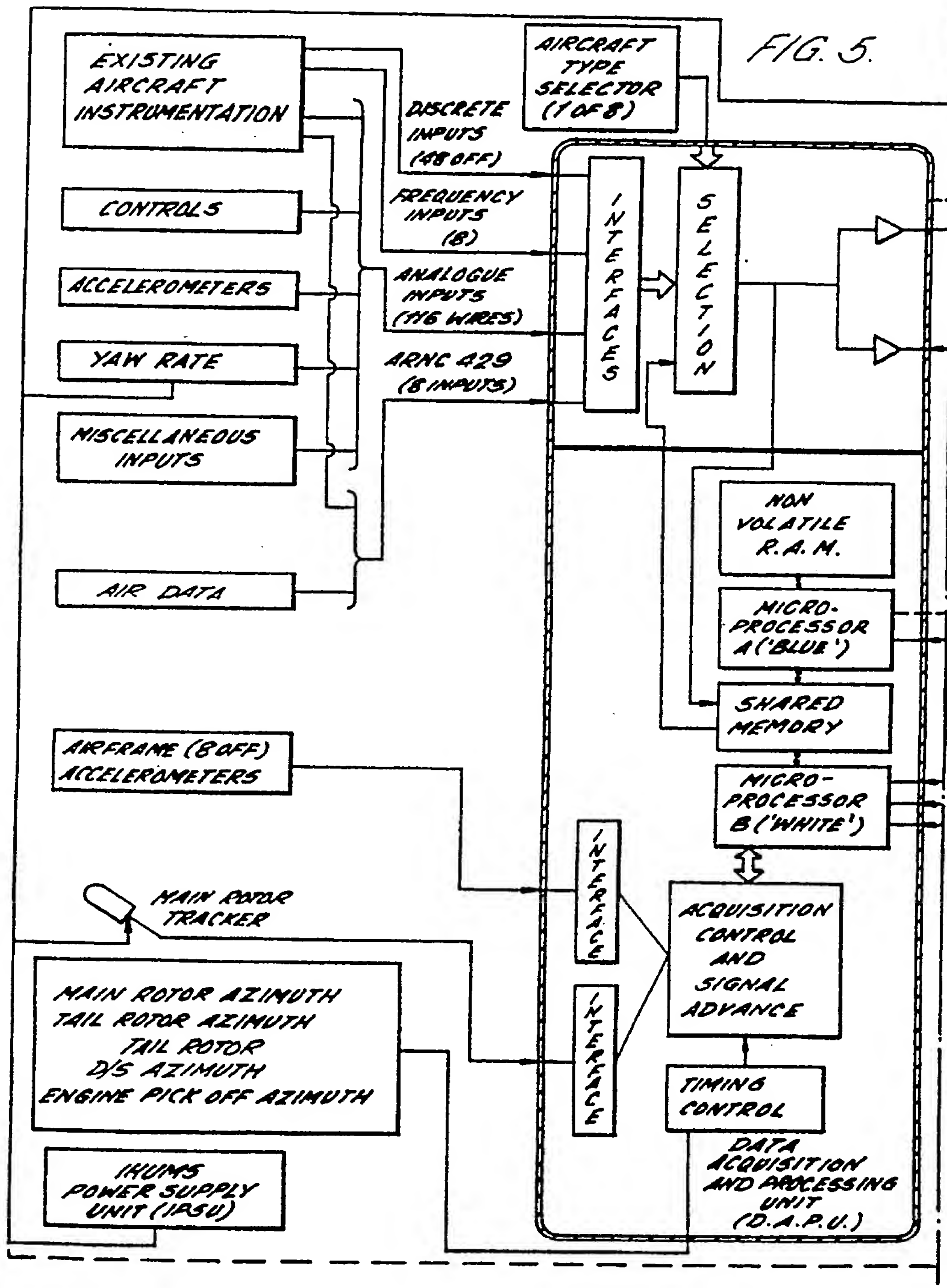
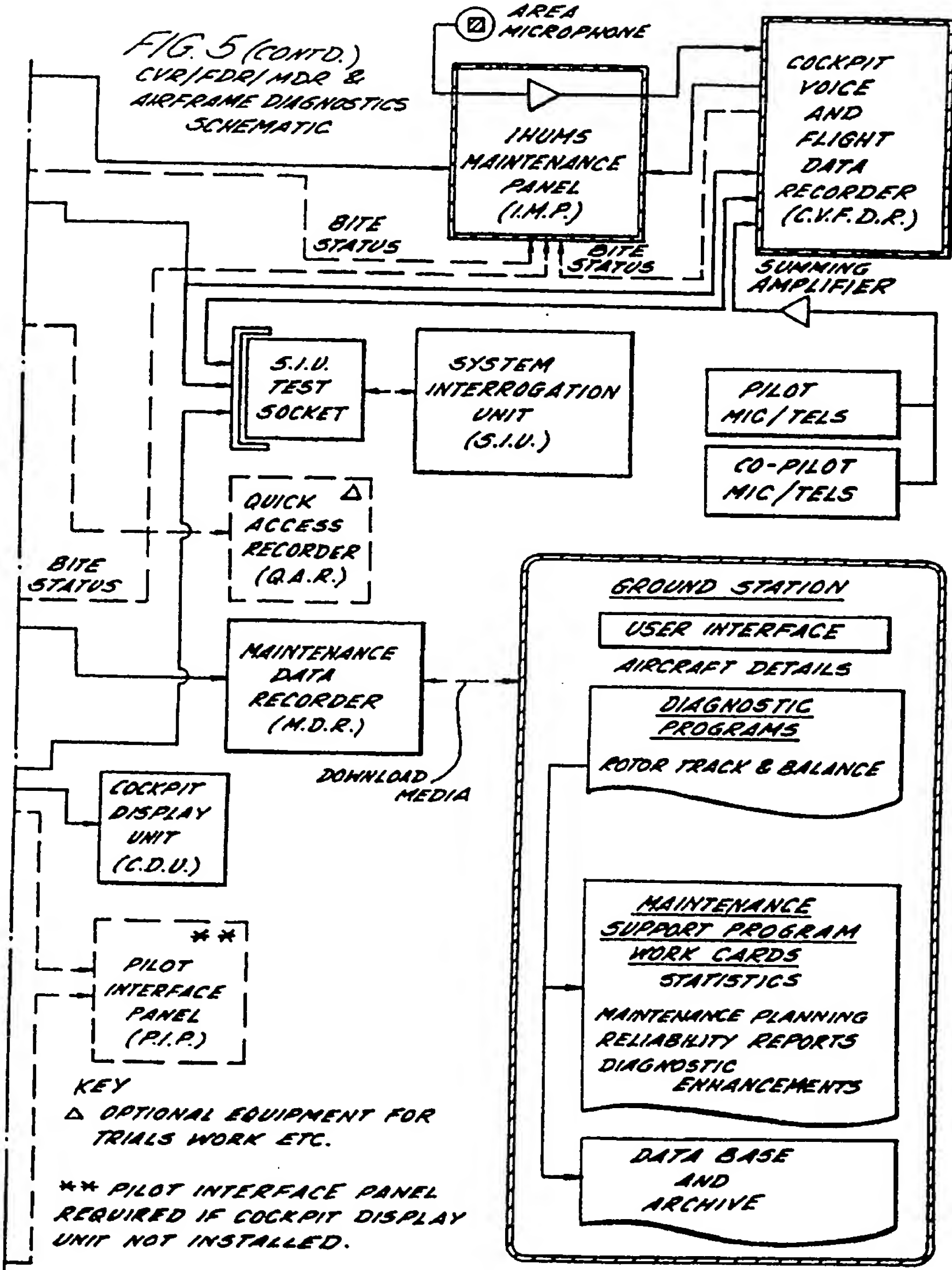


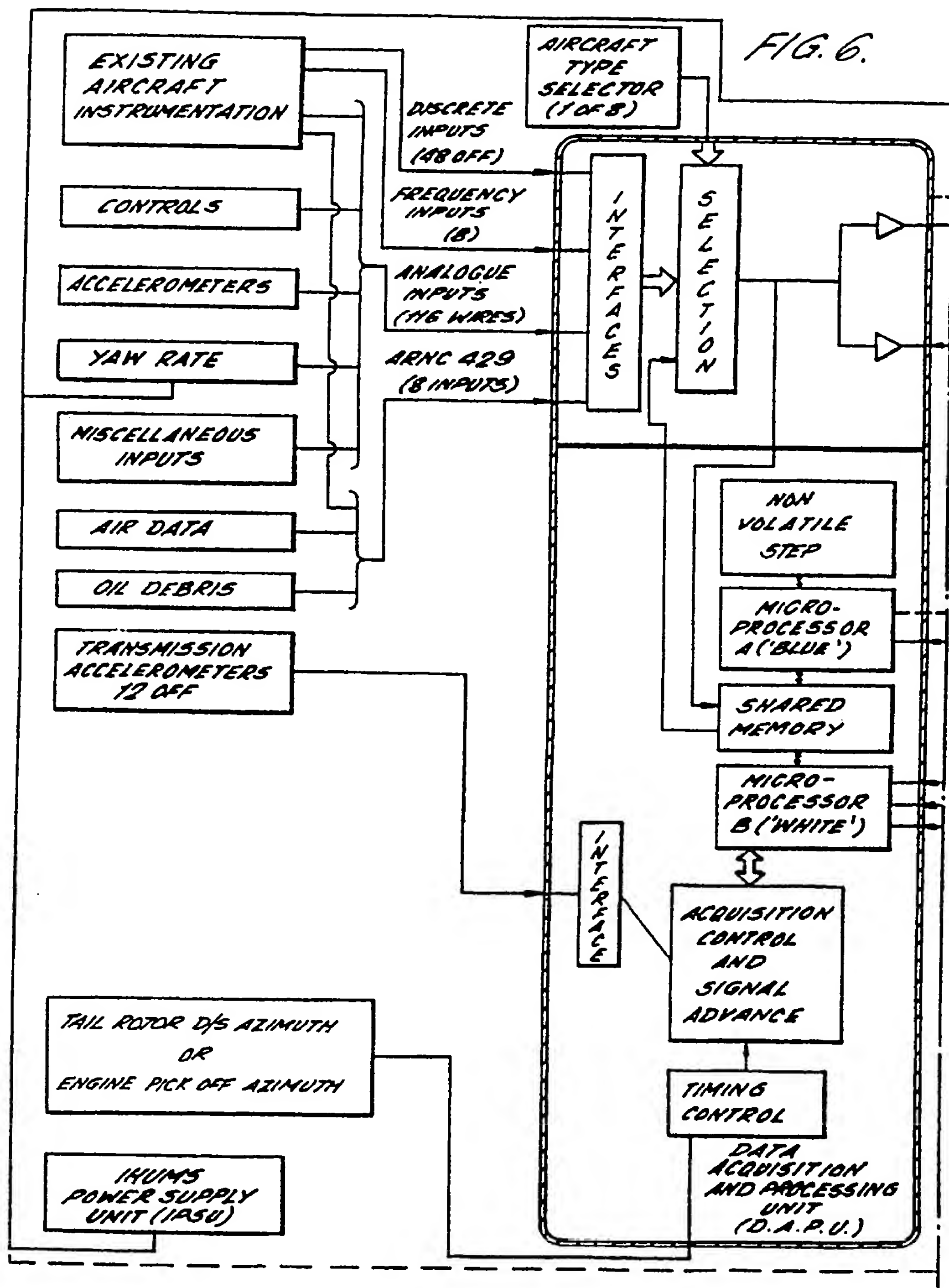
FIG. 4 (CONTD.)  
CVR/FDR/MDR  
SCHEMATIC

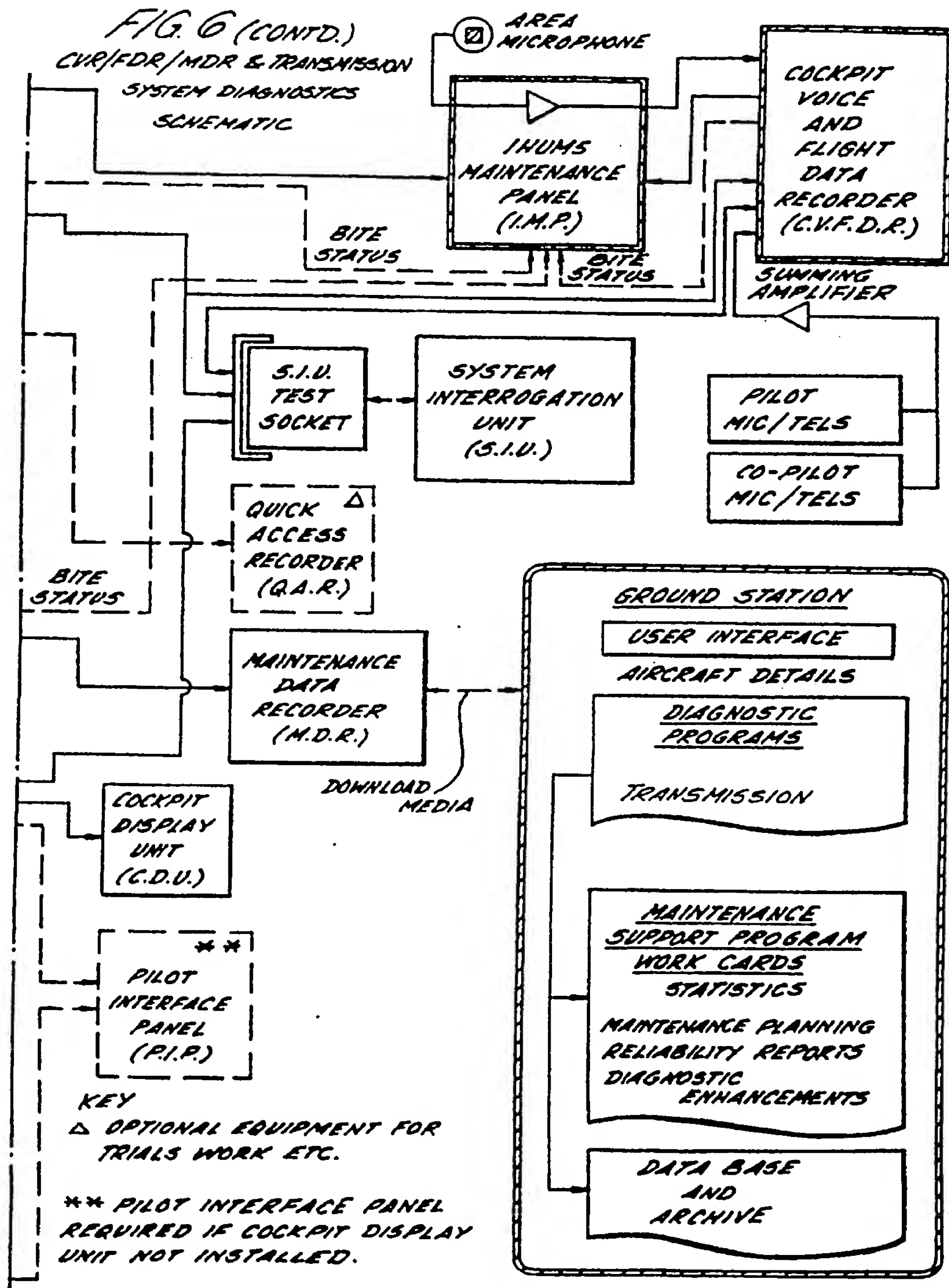


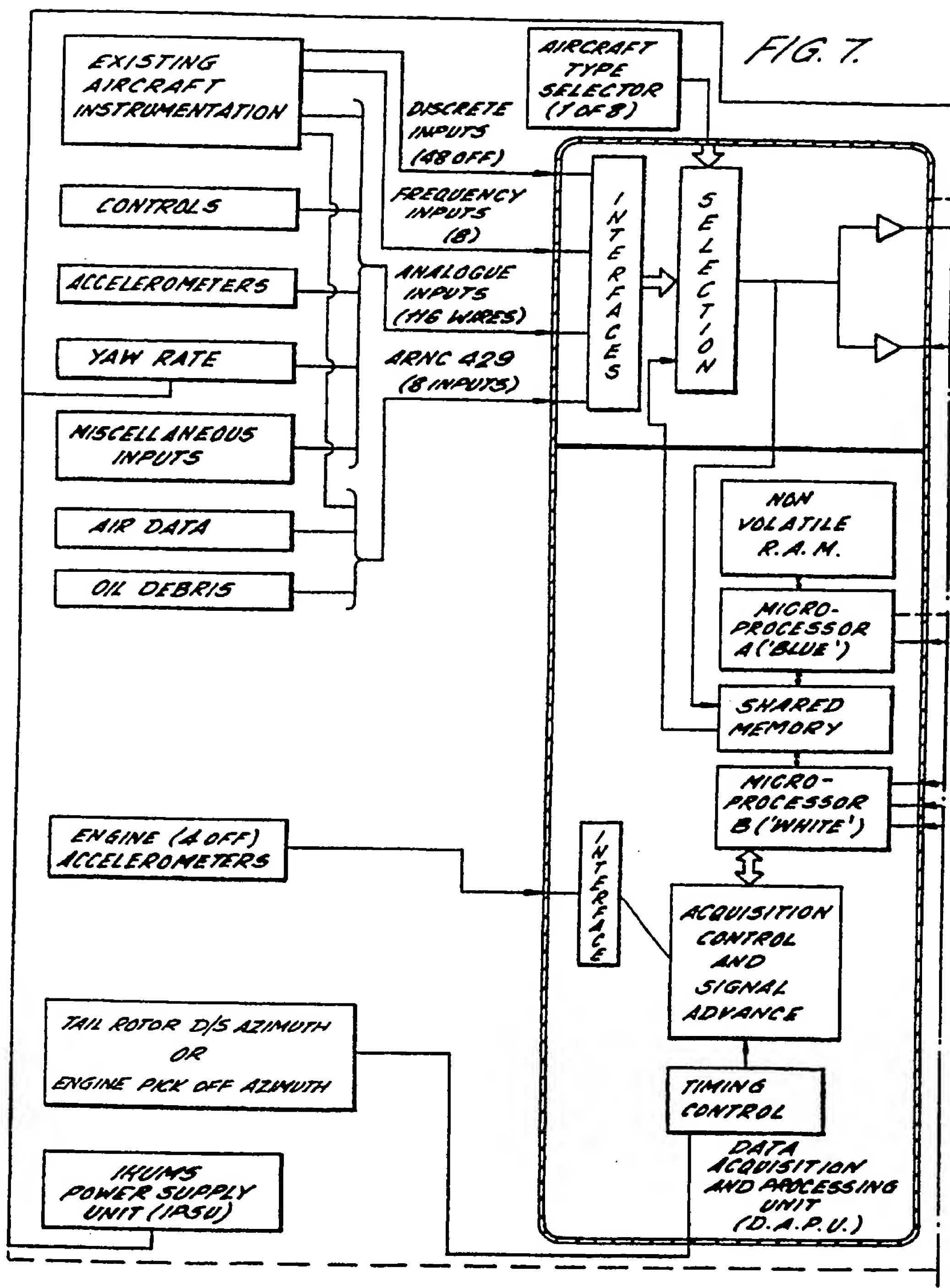












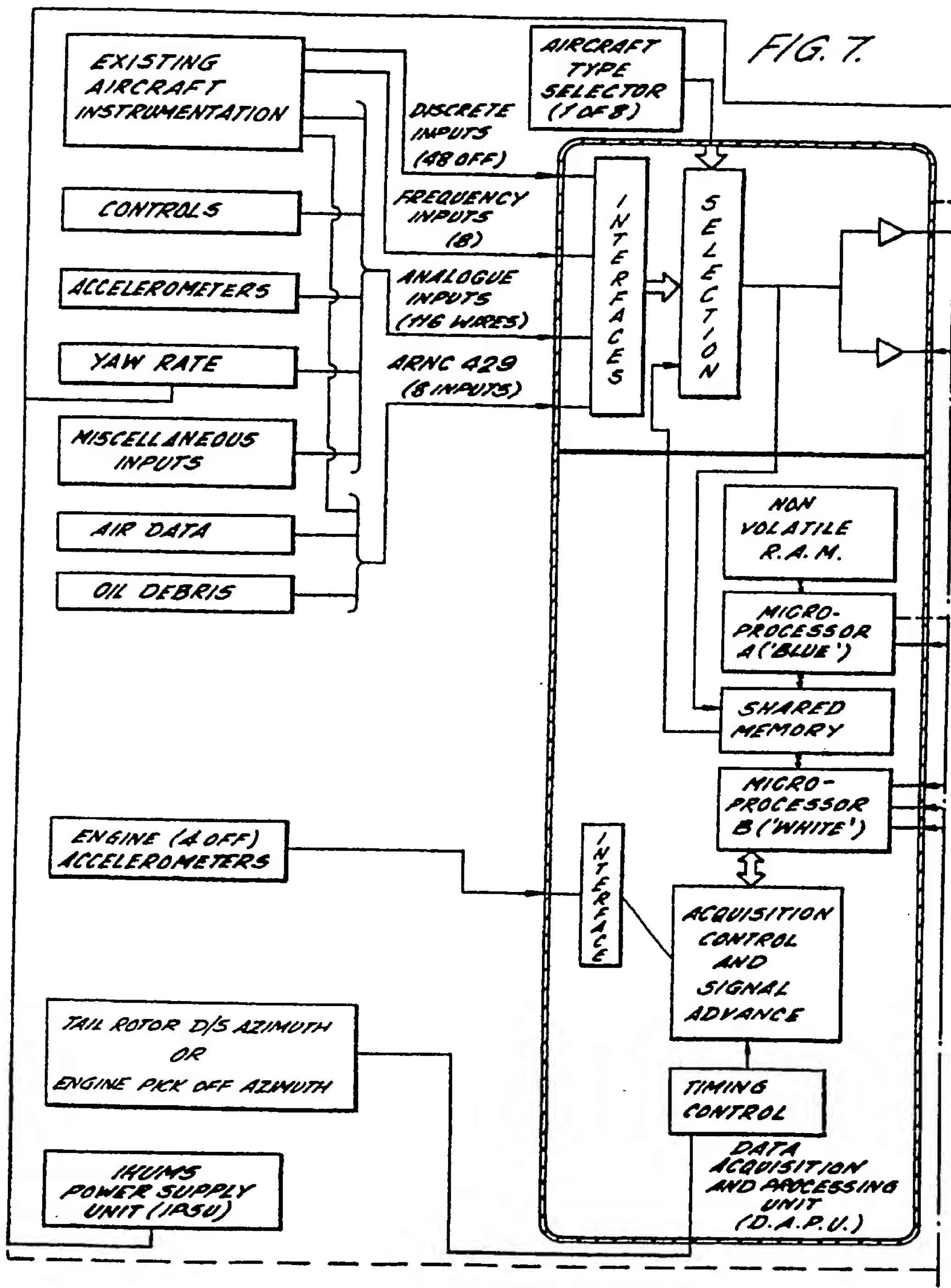


FIG. 8.

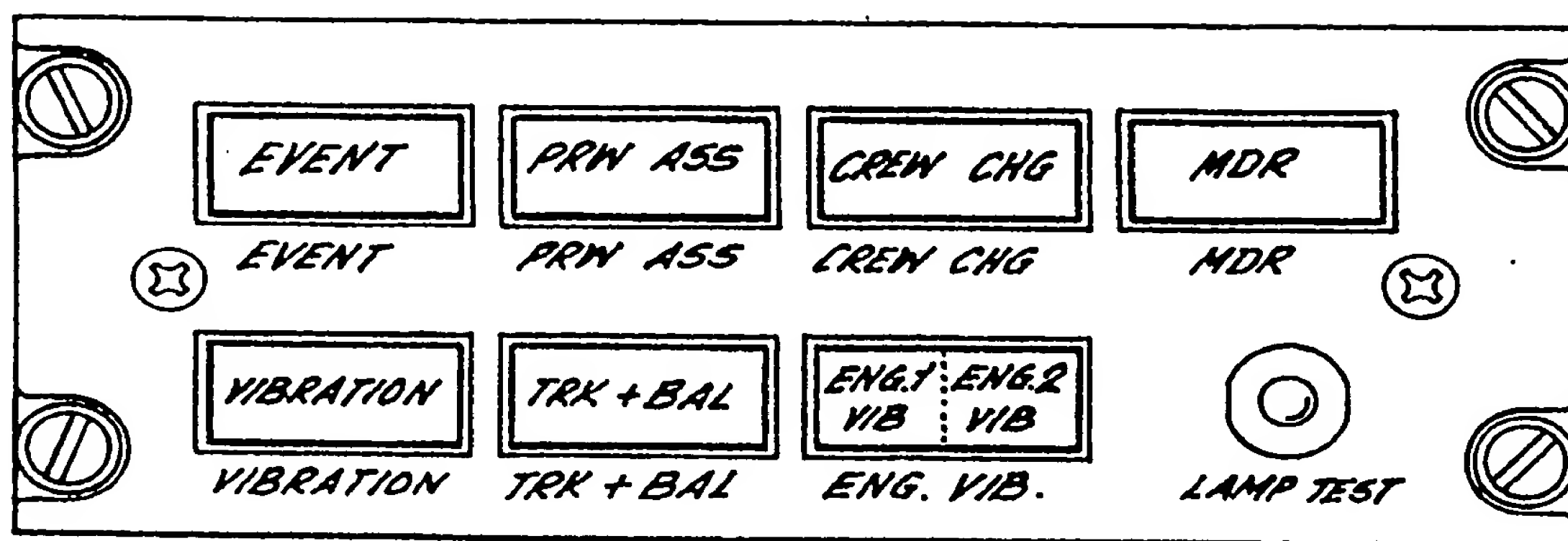




FIG. 9.

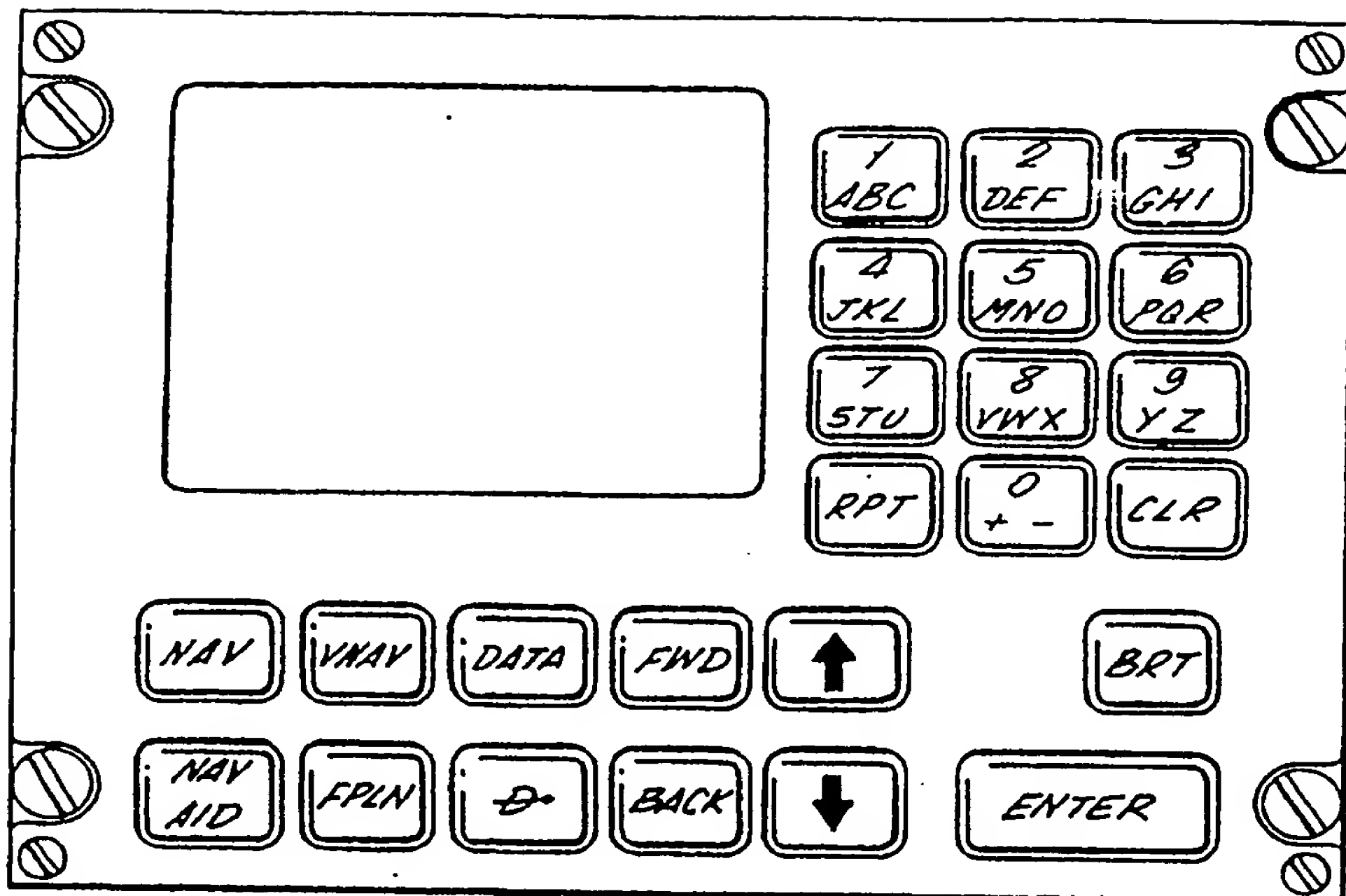
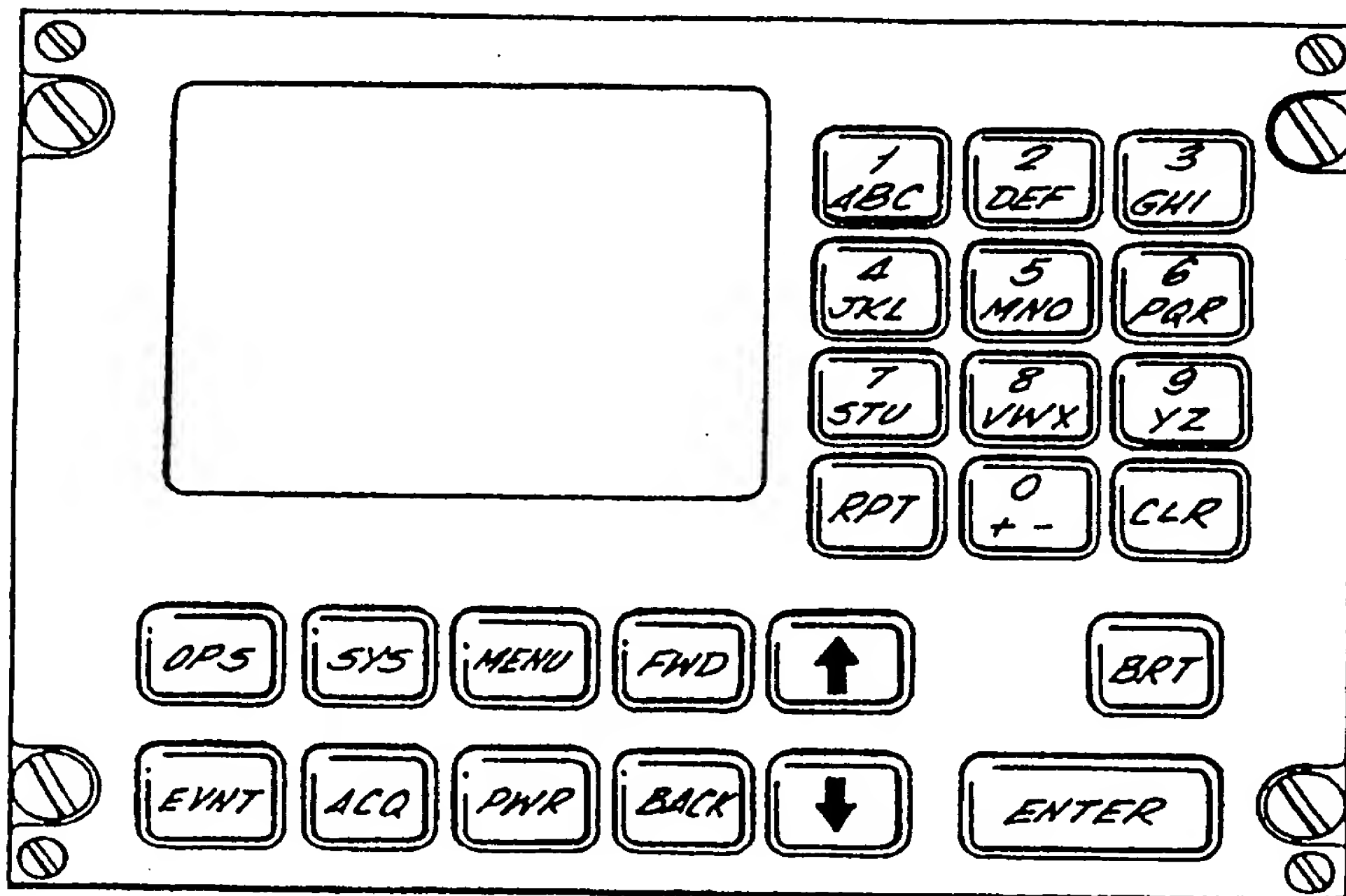


FIG. 10.



*FIG. 11.*  
*THE CDU.*

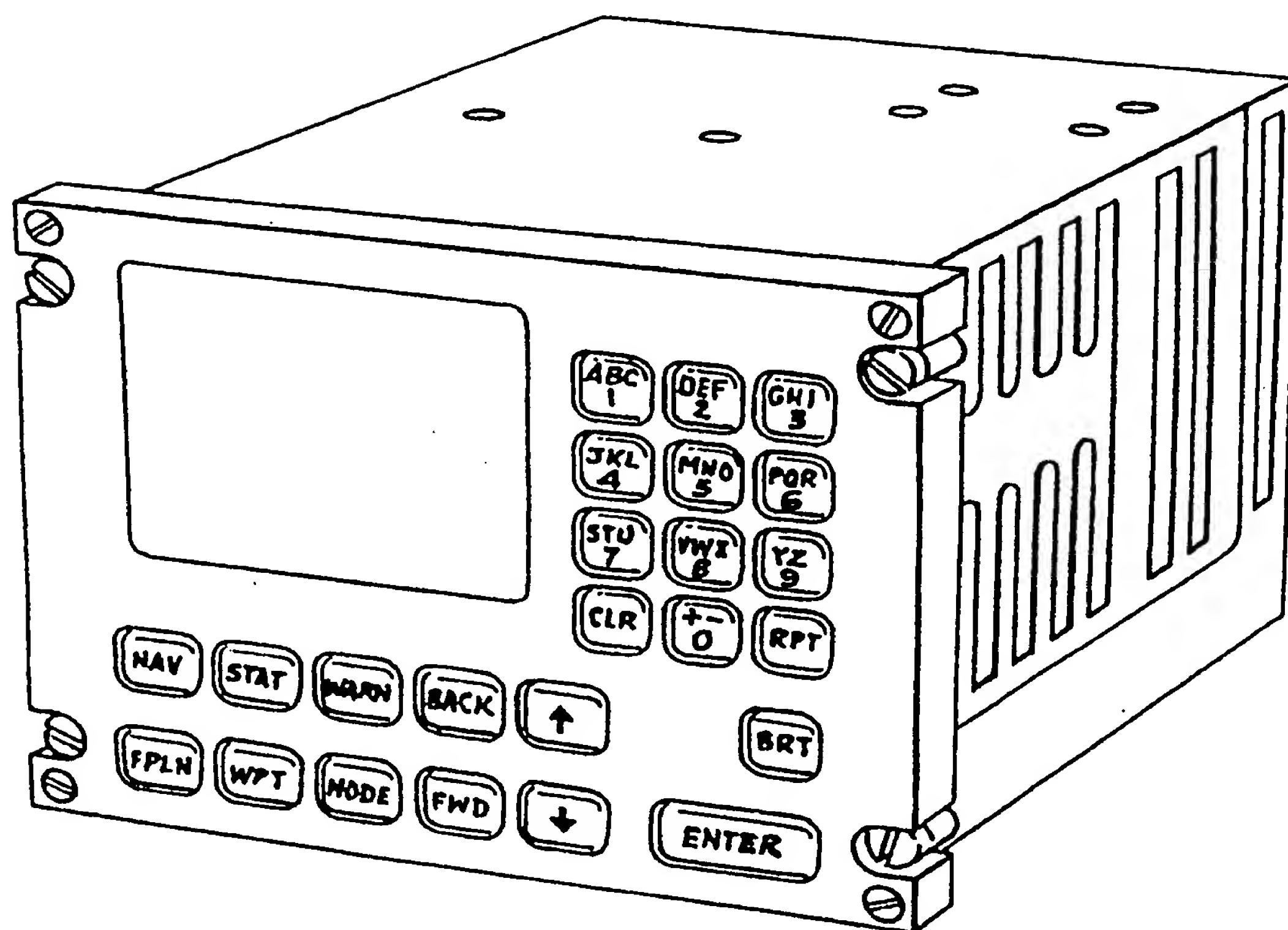


FIG. 13.  
THE P.I.P. PANEL

